
CALIBRATION

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MODIS Science Team Meeting, Oct. 1 - 3, 1991. Attachment CC

CALIBRATION TOPICS

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- CALIBRATION REQUIREMENTS
- CALIBRATION STRATEGY
- OBC
 - SOLAR ILLUMINATED DIFFUSER
 - BLACKBODY
 - SRCA
 - MODELING RESULTS
- PREFLIGHT
 - PERFORMANCE/CHARACTERIZATION
 - BTE/GSE
 - CALIBRATION
- REVIEW OF OTHER MAJOR ASPECTS
 - SPECTRAL SHIFTS
 - DIFFUSER SELECTION
 - LUNAR CALIBRATION
- CALIBRATION SUMMARY

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CALIBRATION REQUIREMENTS

| Parameter | Phase C/D Requirement | Predicted | |
|--|---|--------------------|----------|
| | | Preflight | On-Orbit |
| Radiometric Calibration | | | |
| Below 1.0 μm | 5% | 4% | 3%** |
| 1.1 to 3.0 μm | 5% | 4% | 3%** |
| Above 3 μm | 1% | 1% | 1% |
| Reflectance | 2% | 4% | 2% |
| Spectral Calibration | | | |
| Center Wave-length | 0.5 nm preflight 1.0 nm on-orbit | 0.5 nm | 1.0 nm* |
| Spectral Band-to-Band Stability | 0.5% FS 1.0% HS | 0.5% FS 1.0% HS | 0.5% FS |
| Geometric Calibration | | | |
| Band-to-Band Registration | 0.1 IFOV | 0.05 IFOV | 0.1 IFOV |
| Diffuser BRDF | | | |
| <2.0 μm | 1.0% | | |
| 2.0 to 2.5 μm | 1.5% | | |
| FS = Full Scale HS = Half Scale | | | |
| * Dependent on good correlation with full aperture ground measurement and SRCA sub-aperture measurements | | | |
| ** Multiple calibration methodologies are required | | | |

CALIBRATION STRATEGY USE MULTIPLE MEASUREMENT/CALIBRATION METHODOLOGIES



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- PREFLIGHT — USE MULTIPLE METHODS AND CORRELATE
- LOOK FOR TRENDS IN CALIBRATION HISTORY THROUGHOUT PREFLIGHT LIFETIME
- INFLIGHT — ONBOARD CALIBRATORS
- RADIOMETRIC, SPATIAL, SPECTRAL AND ONBOARD CALIBRATOR STABILITY MONITORING
- INFLIGHT — REMOTE SCENES
- MULTIPLE EARTH SCENES
 - LUNAR SCENE

EOS SENSOR CALIBRATION

RECOMMENDED COORDINATED CALIBRATION CAMPAIGN COVERING MULTIPLE EOS SENSORS

- PREFLIGHT CALIBRATION SOURCES
- INFLIGHT EARTH REFERENCE SCENES

RESULT

IT IS BELIEVED THAT 2% ABSOLUTE CALIBRATION IS
ACHIEVABLE IF THE CONCEPT OF MULTIPLE
MEASUREMENT/CALIBRATION METHODOLOGIES IS FULLY
IMPLEMENTED

MULTIPLE OBC ARE FEASIBLE AND WORKABLE

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DC RESTORE

ALL BANDS

BLACKBODY — FULL APERTURE

MWIR/LWIR

SOLAR ILLUMINATED DIFFUSER
TWO LEVELS

VIS/NIR/SWIR

~~POLARIZATION CALIBRATOR~~

~~BANDS 17-19~~

SRCA

RADIOMETRIC CALIBRATION CHECK

VIS/NIR/SWIR

SPATIAL BAND-TO-BAND REGISTRATION

VIS/NIR/SWIR/MWIR/LWIR

SPECTRAL RESPONSE

VIS/NIR/SWIR

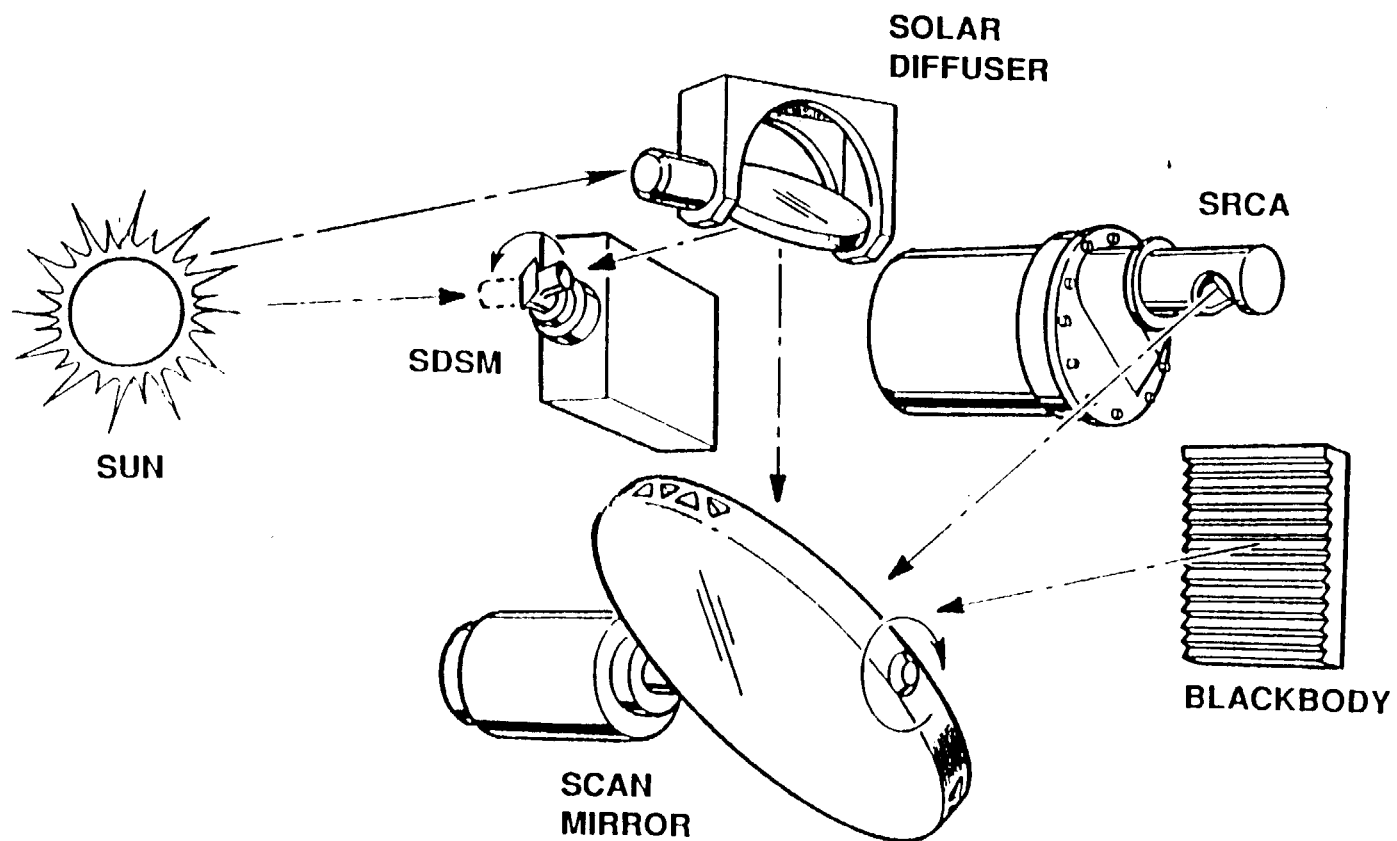
DIFFUSER STABILITY MONITOR

VIS/NIR

MODIS-N IN-FLIGHT CALIBRATORS

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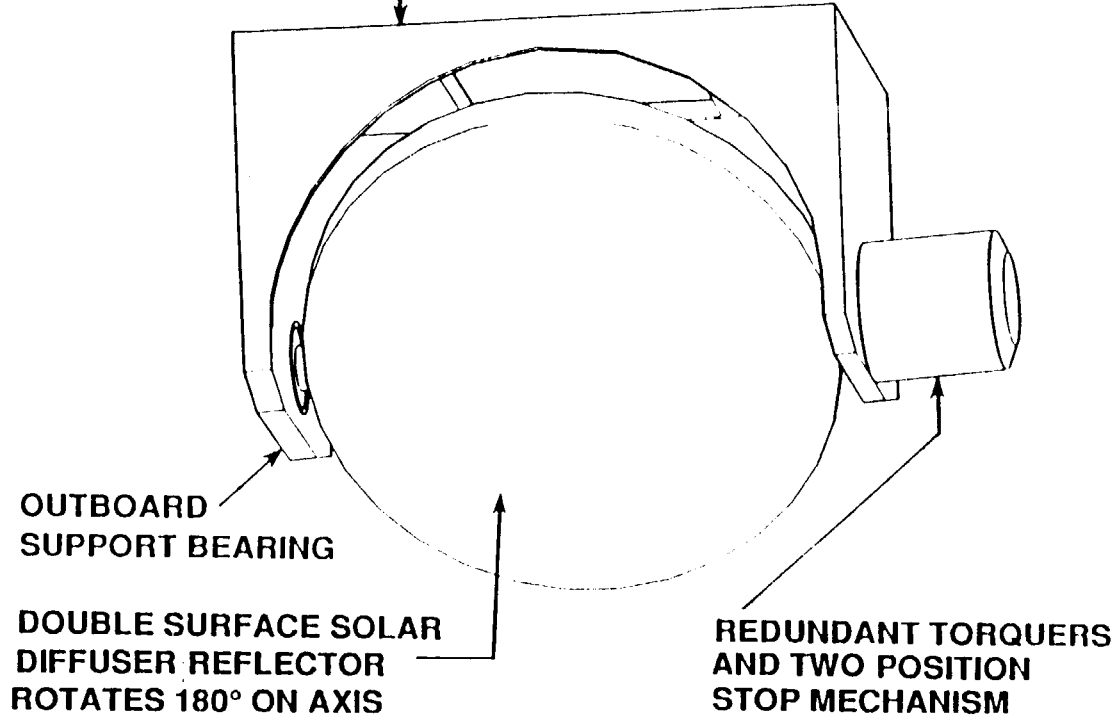
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MAIN HOUSING BOLTS TO
TOP OF SENSOR HOUSING



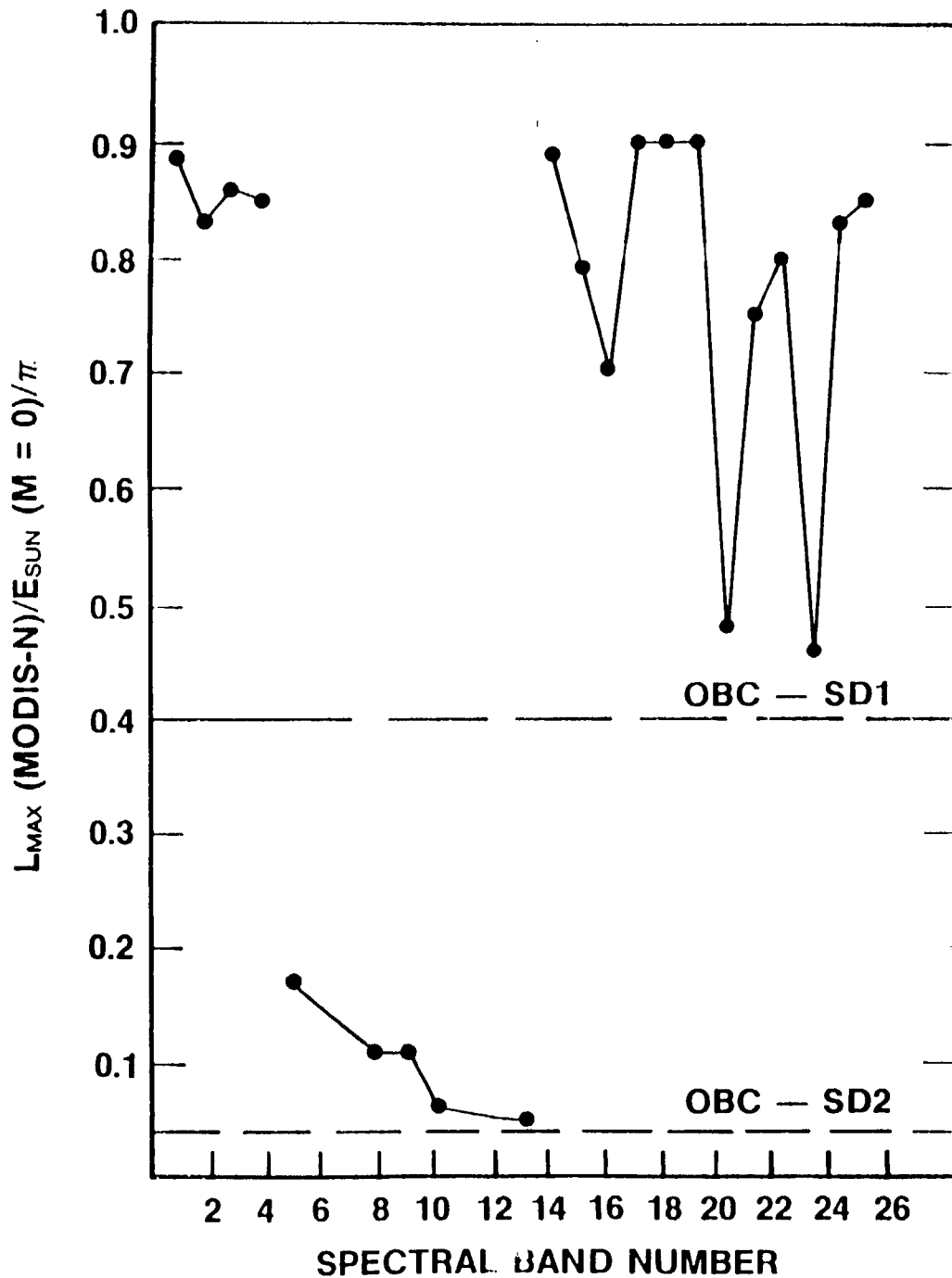
**FULL APERTURE
SOLAR DIFFUSER
PROVIDES
ACCURATE
AND STABLE
RADIOMETRIC
CALIBRATION**

- **DOUBLE SIDED**
- **BRDF's: 0.18/sr, 0.018/sr**
- **EFFECTIVE ALBEDO: 46%, 4.6%**



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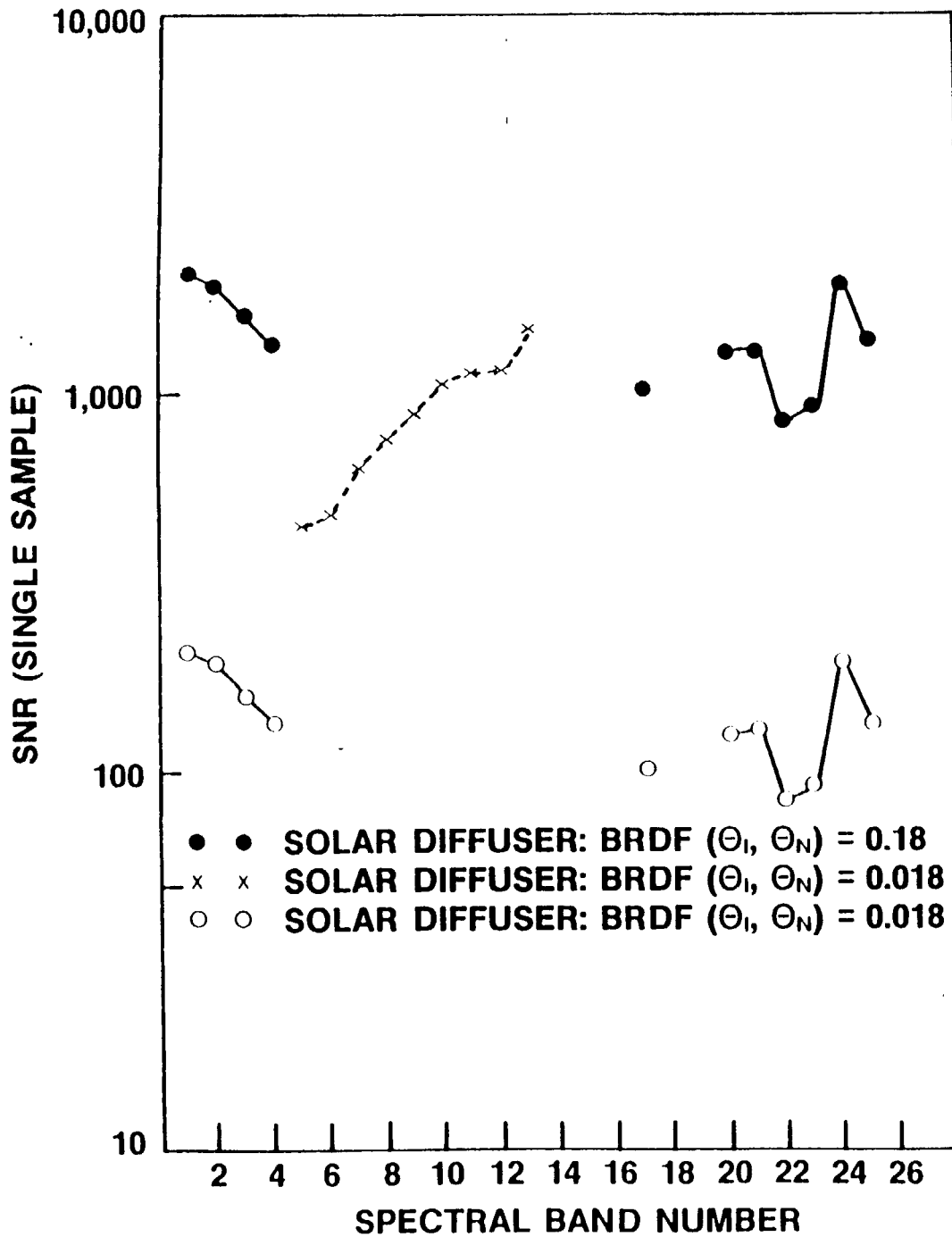
TWO SOLAR DIFFUSERS NEEDED TO MATCH VIS/NIR AND SWIR ALBEDO LEVELS





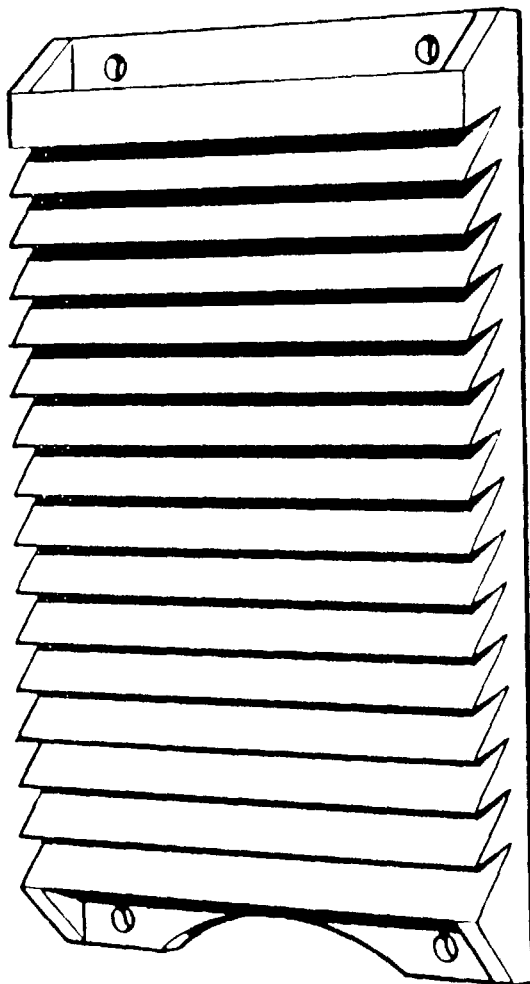
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TWO SOLAR DIFFUSERS PERMIT ADEQUATE SNRS FOR RADIOMETRIC CALIBRATION



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- 25° V-GROVE HALF ANGLE
- ALUMINUM STRUCTURE
- AMBIENT TEMPERATURE
- EMISSIVITY ≥ 0.992

**V-GROVE
BLACKBODY
DESIGN
OFFERS HIGH
EMISSIVITY**

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ON ORBIT RADIOMETRIC CALIBRATION MODEL FOR MWIR/LWIR

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- **PARAMETRIC MODELING — MODEL VARIABLES**

- SCENE - TEMPERATURE
- SENSOR - SPECTRAL SHIFT, OUT OF BAND TRANSMITTANCE, $NE\Delta T$
- OBC-BB - TEMPERATURE AND EMISSIVITY AND THEIR UNCERTAINTY
- SPURIOUS SOURCES - DIRECT AND INDIRECT SOLAR ILLUMINATION, SENSOR INTERIOR CAVITY EMISSION

- **MODEL SNAPSHOT**

- SCENE - NOMINAL TEMPERATURE FOR EACH BAND
- SENSOR - SPECTRAL SHIFT 0.01 TO 0.05 μm , $T_{oob} = 0.001$, SOW $NE\Delta T$ SET
- OBC-BB - $T_{BB} = 320K$; $\epsilon_{BB} = 0.99$
 $T_{BB_UNC} = 0.25K$; $\epsilon_{BB_UNC} = 0.006$
- SPURIOUS SOURCES - NO DIRECT SUNLIGHT: INDIRECT $\rho = 0.06$ WITH $\Omega = 1$ sr;
SENSOR CAVITY: $\epsilon = 1.0$, $T = 300K$, $\Omega = 2$ sr

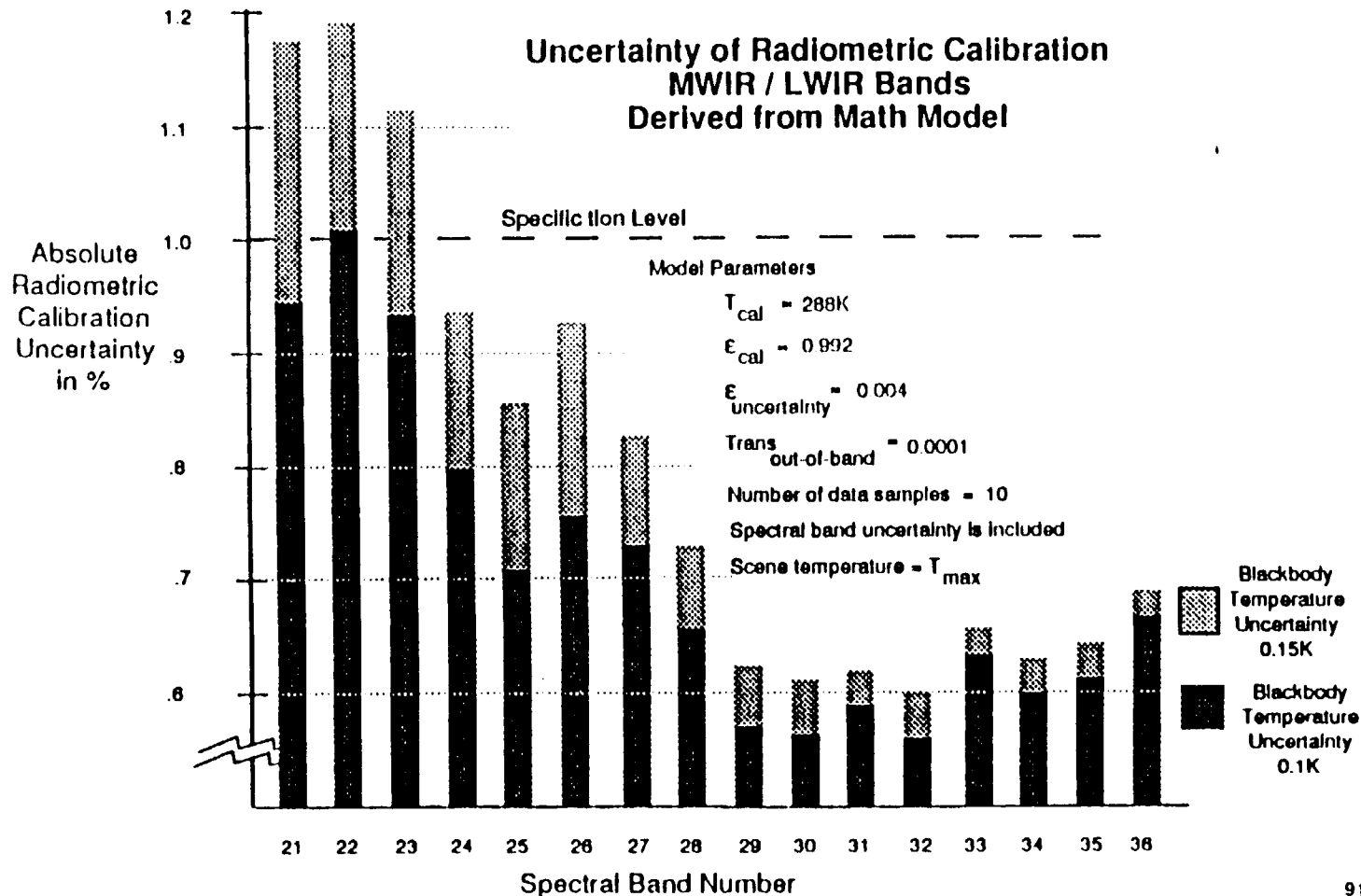
- **TEMPERATURE UNCERTAINTY:**

- RSS-0.30K TO 0.56K

ACCURACY PREDICTION FOR MWIR AND LWIR BANDS

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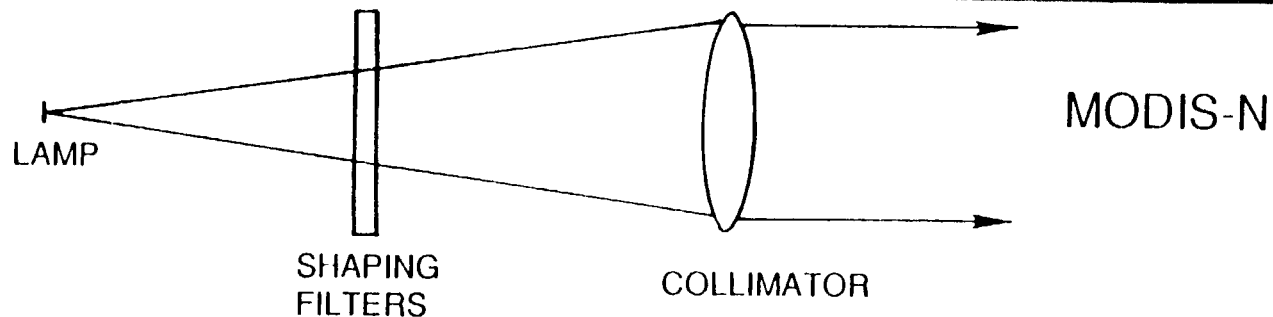
SRCA — PRIMARY FUNCTIONS

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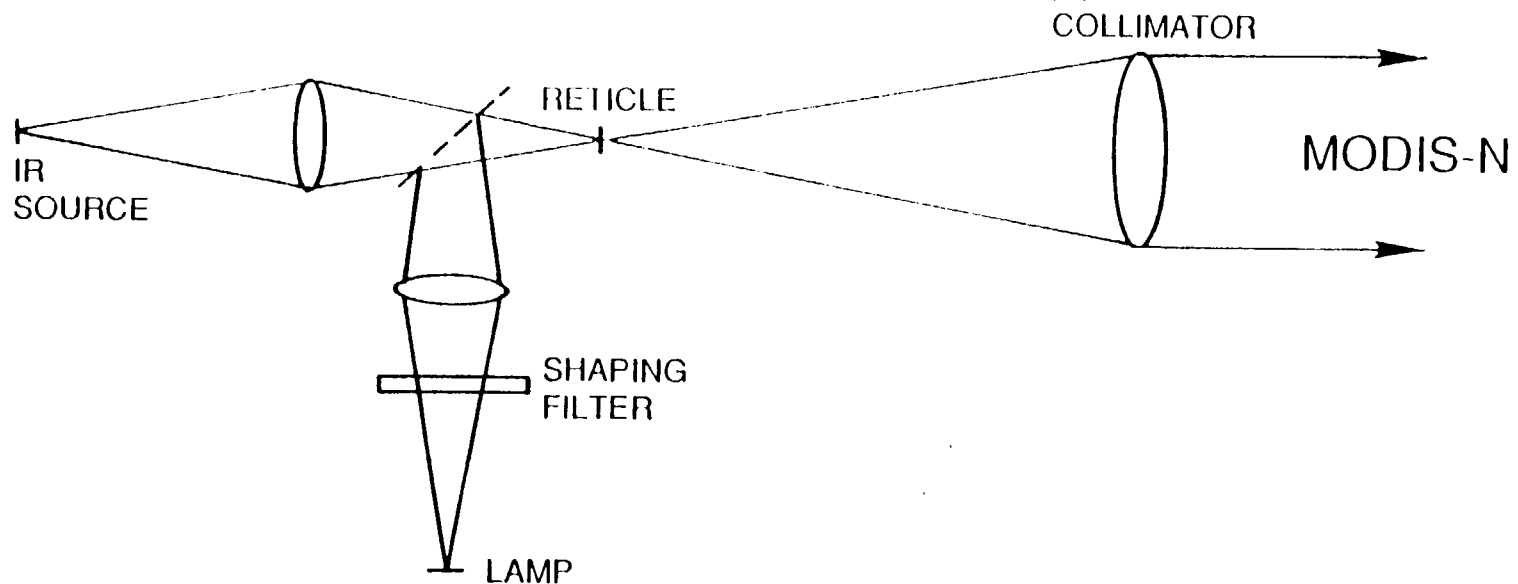
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- **RADIOMETRIC CALIBRATION CHECK** **VIS/NIR/SWIR**
- **SPATIAL BAND-TO-BAND REGISTRATION** **VIS/NIR/SWIR/
MWIR/LWIR**
- **MONOCHROMATOR SPECTRAL CALIBRATOR** **VIS/NIR/SWIR**
- **SOLAR ILLUMINATED DIFFUSER STABILITY MONITOR** **VIS/NIR**

CONCEPTUAL SIMPLICITY OF INDIVIDUAL SRCA FUNCTIONS (1)

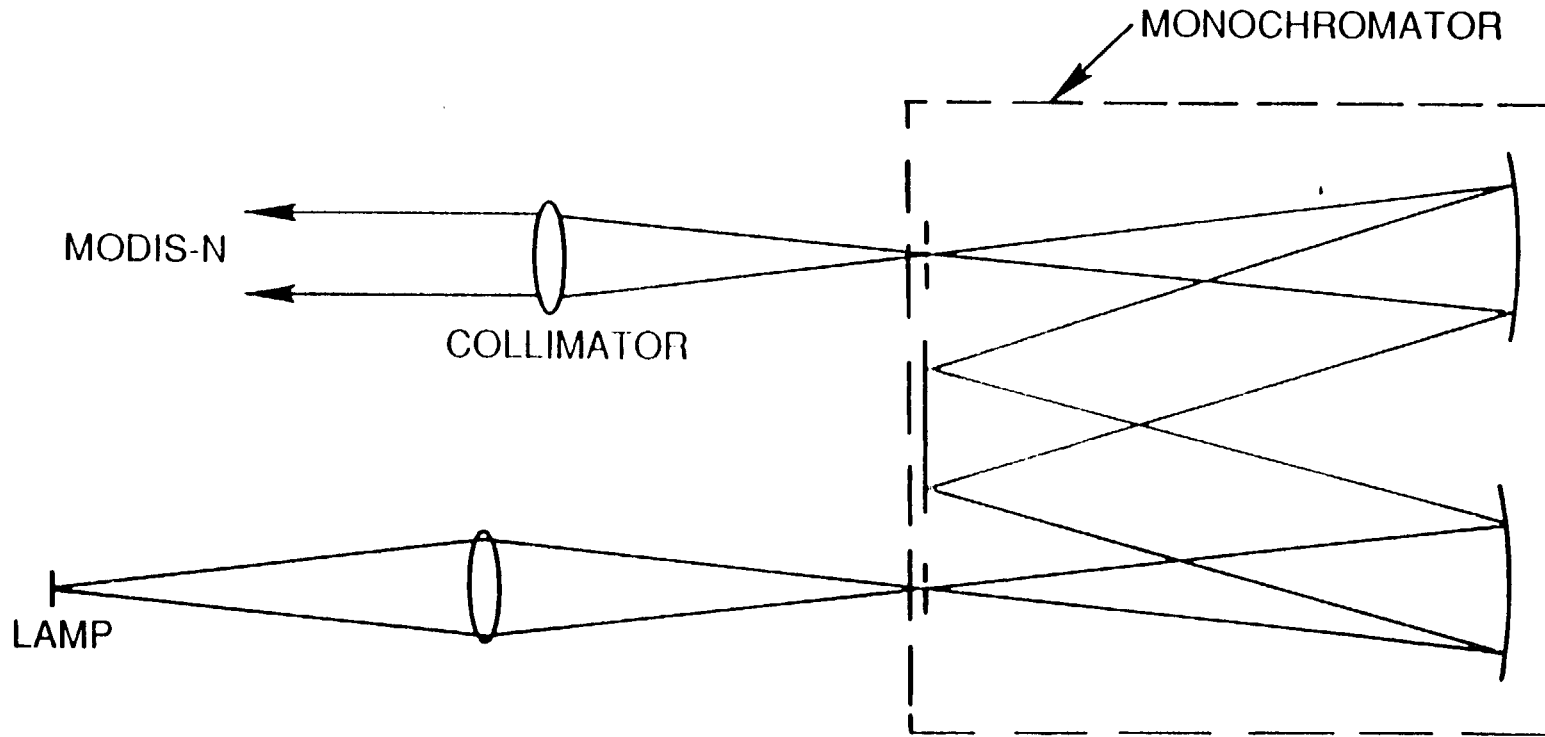


RADIOMETRIC CALIBRATION CHECK



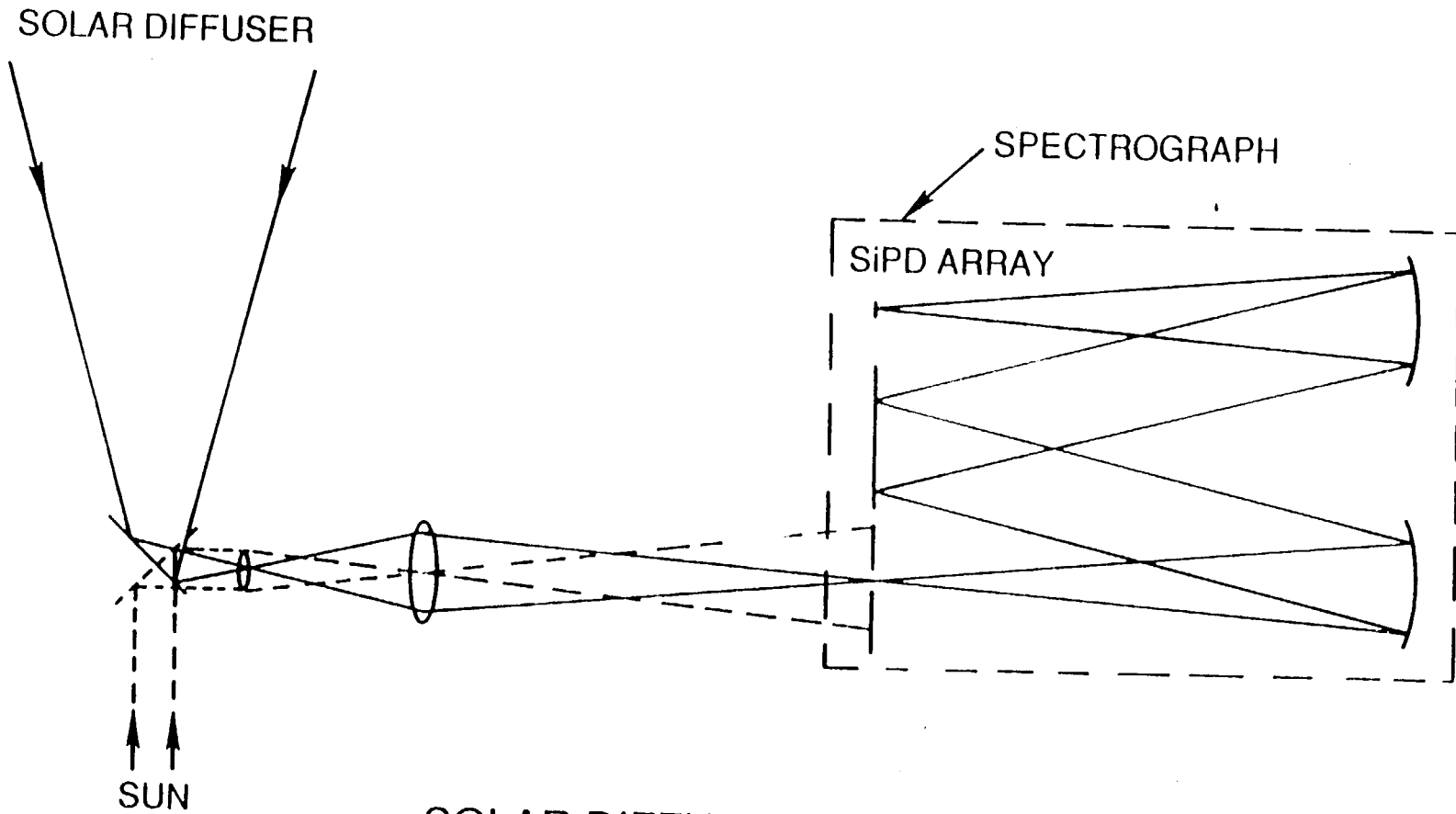
SPATIAL BAND TO BAND REGISTRATION

CONCEPTUAL SIMPLICITY OF INDIVIDUAL SRCA FUNCTIONS (2)



MONOCHROMATOR SPECTRAL CALIBRATOR

CONCEPTUAL SIMPLICITY OF INDIVIDUAL SRCA FUNCTIONS (3)

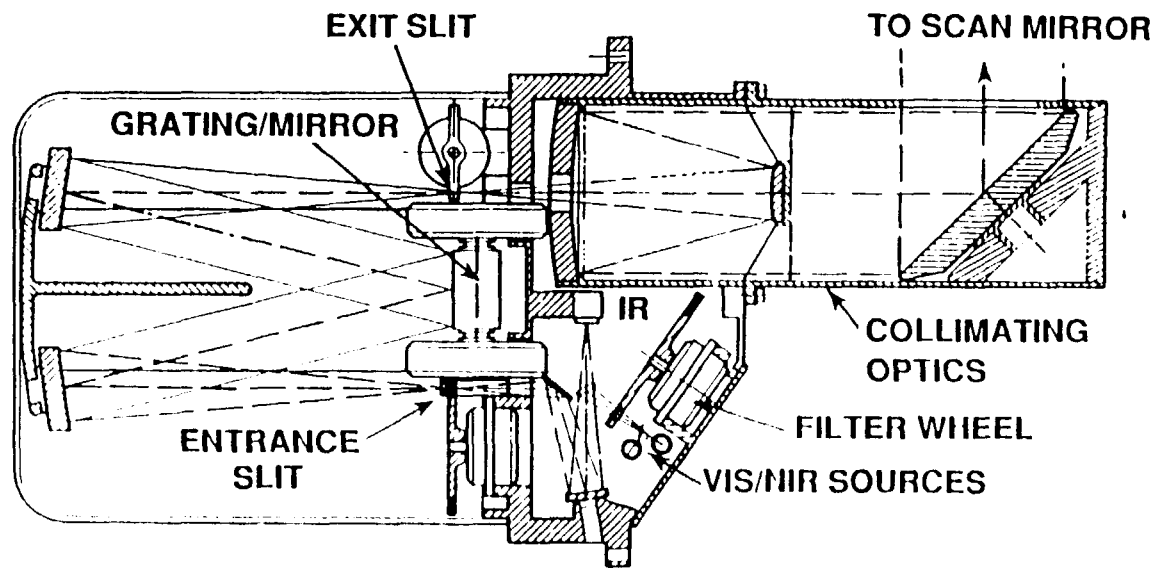


SOLAR DIFFUSER STABILITY MONITOR

SPECTRORADIOMETRIC CALIBRATION ASSEMBLY (SRCA)



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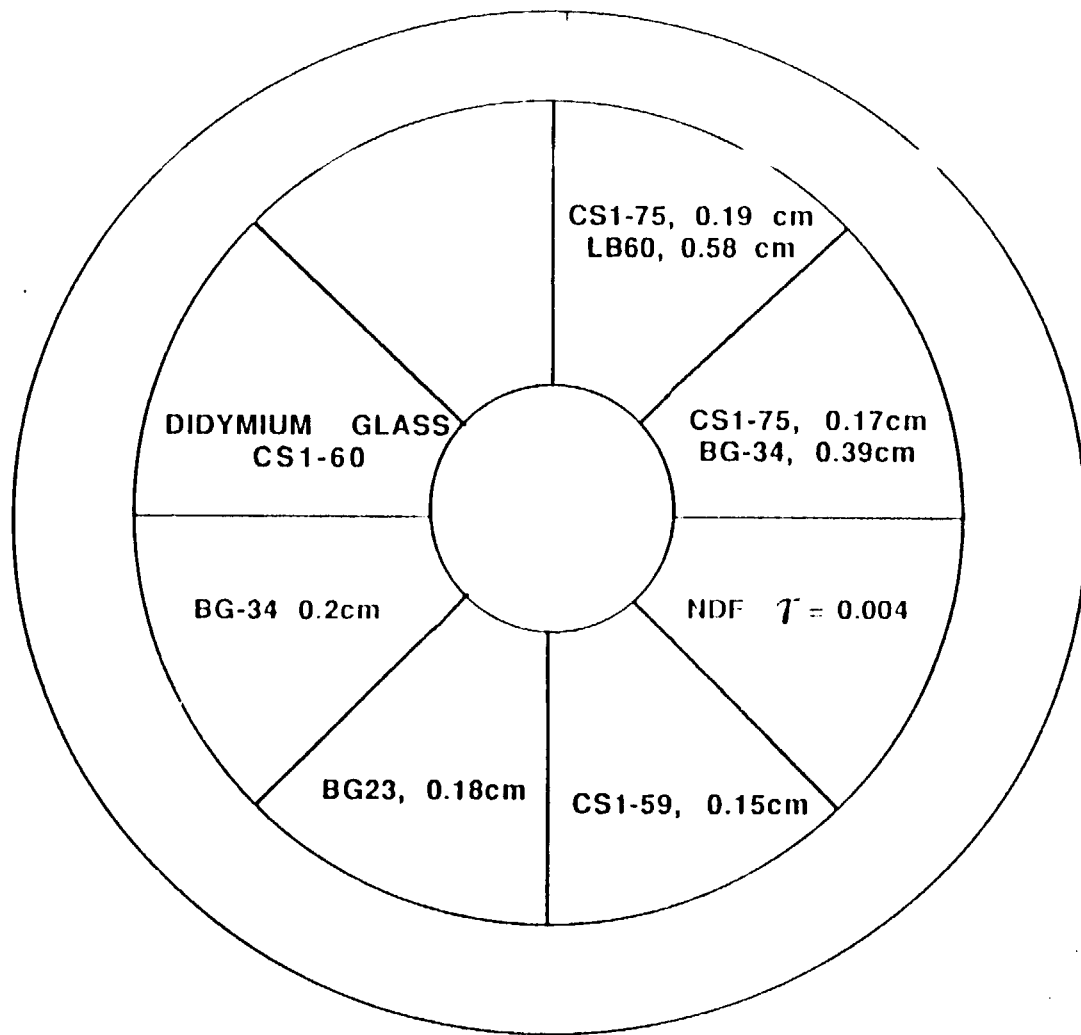


SRCA

| Operating Mode | Sources | Filter Wheel | Relay Optics | Entrance Slit | Collimating Optics | Grating/Mirror | Focusing Optics | Exit Slit | Collimating Optics | Scan Mirror Assembly | Telescope/Aft Optics | Filters/Detectors |
|----------------------|--------------|--------------------------|--------------|-------------------------|--------------------|----------------|-----------------|------------------------|--------------------|----------------------|----------------------|-------------------|
| Radiometric Check | VIS/NIR | | X | Open | X | Mirror | X | Open | X | X | X | X |
| Spectral Calibration | VIS/NIR | Spectral Shaping Filters | X | Entrance Slit / Filters | X | Grating | X | Exit Slit / Photodiode | X | X | X | X |
| Spatial Registration | VIS/NIR & IR | | X | Open | X | Mirror | X | Reticle Pattern | X | X | X | X |
| Self Calibration | VIS/NIR | Duymum Glass | X | Entrance Slit / Filters | X | Grating | X | Photodiode / Exit Slit | X | | | |



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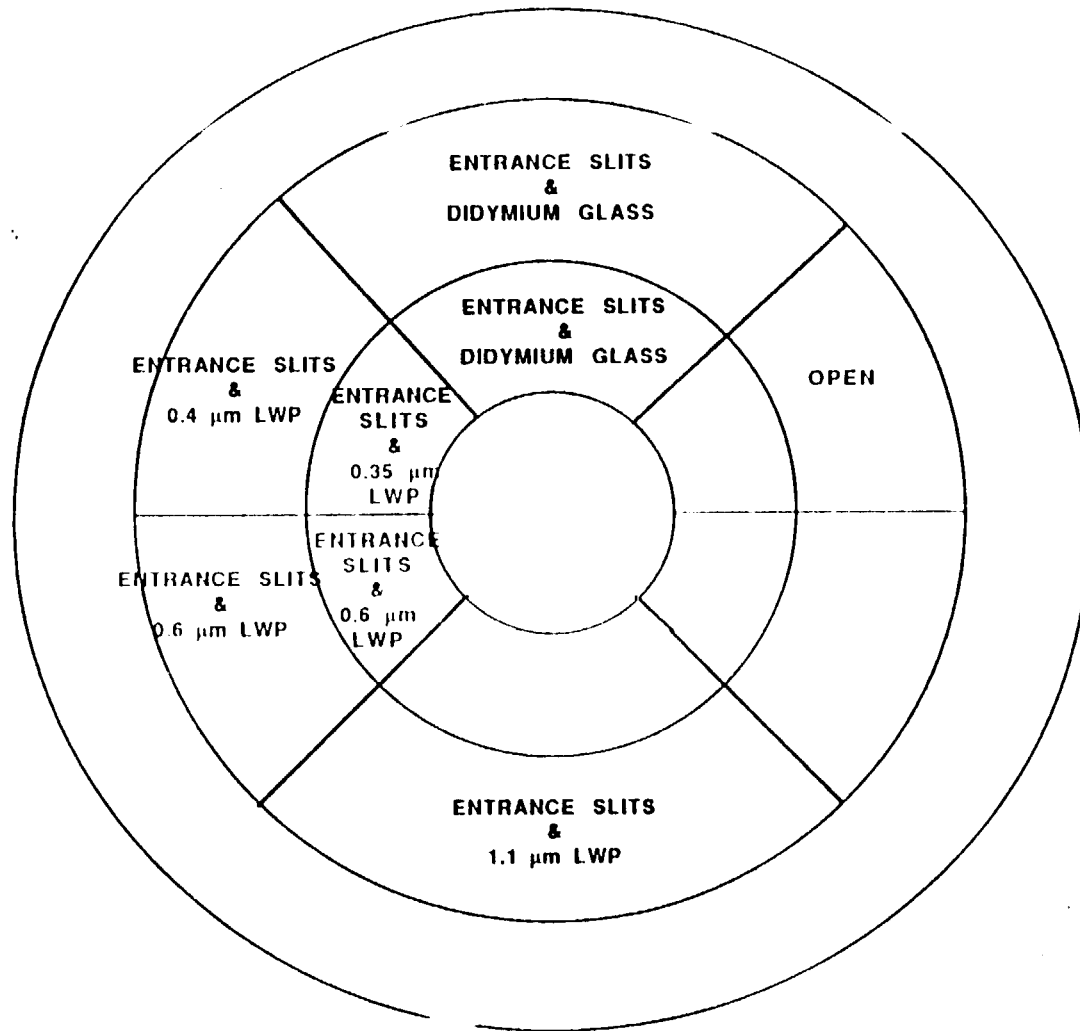


VIS/NIR/SWIR SOURCE ASSEMBLY- FILTER WHEEL

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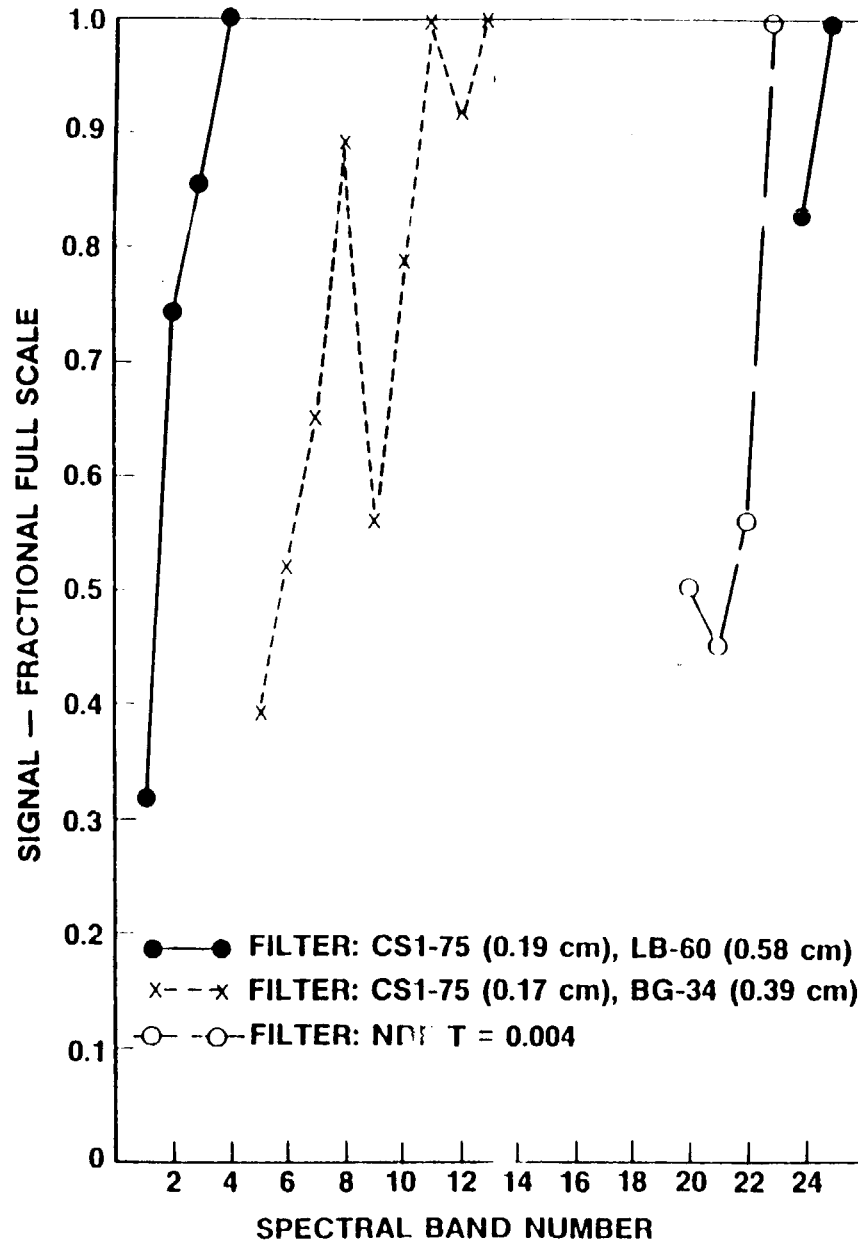


ENTRANCE PLANE MECHANISM

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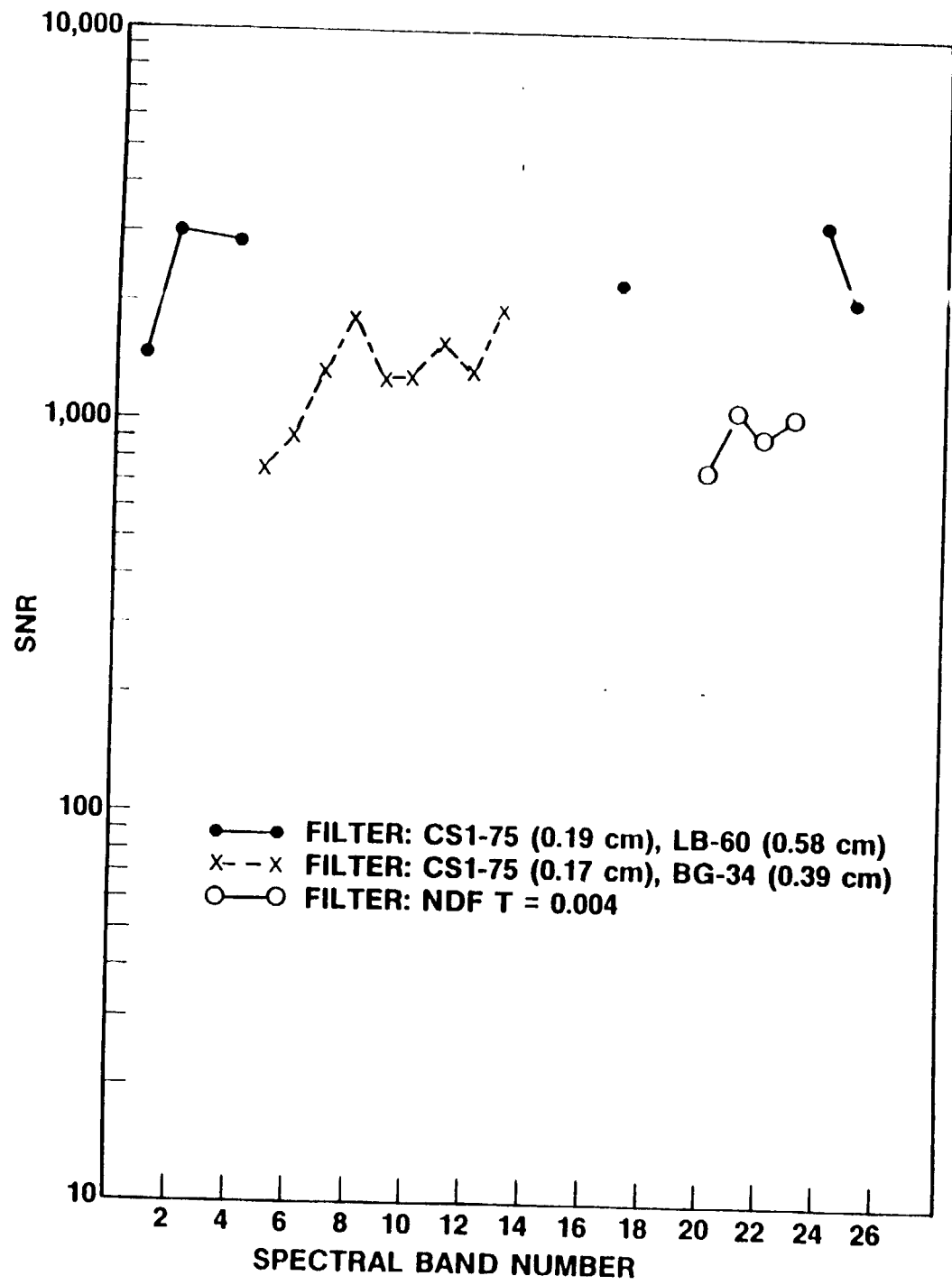
FILTERS SHAPE SRCA RADIANCE FOR ON-BOARD RADIOMETRIC CALIBRATION AND SPATIAL REGISTRATION VIS/NIR/SWIR

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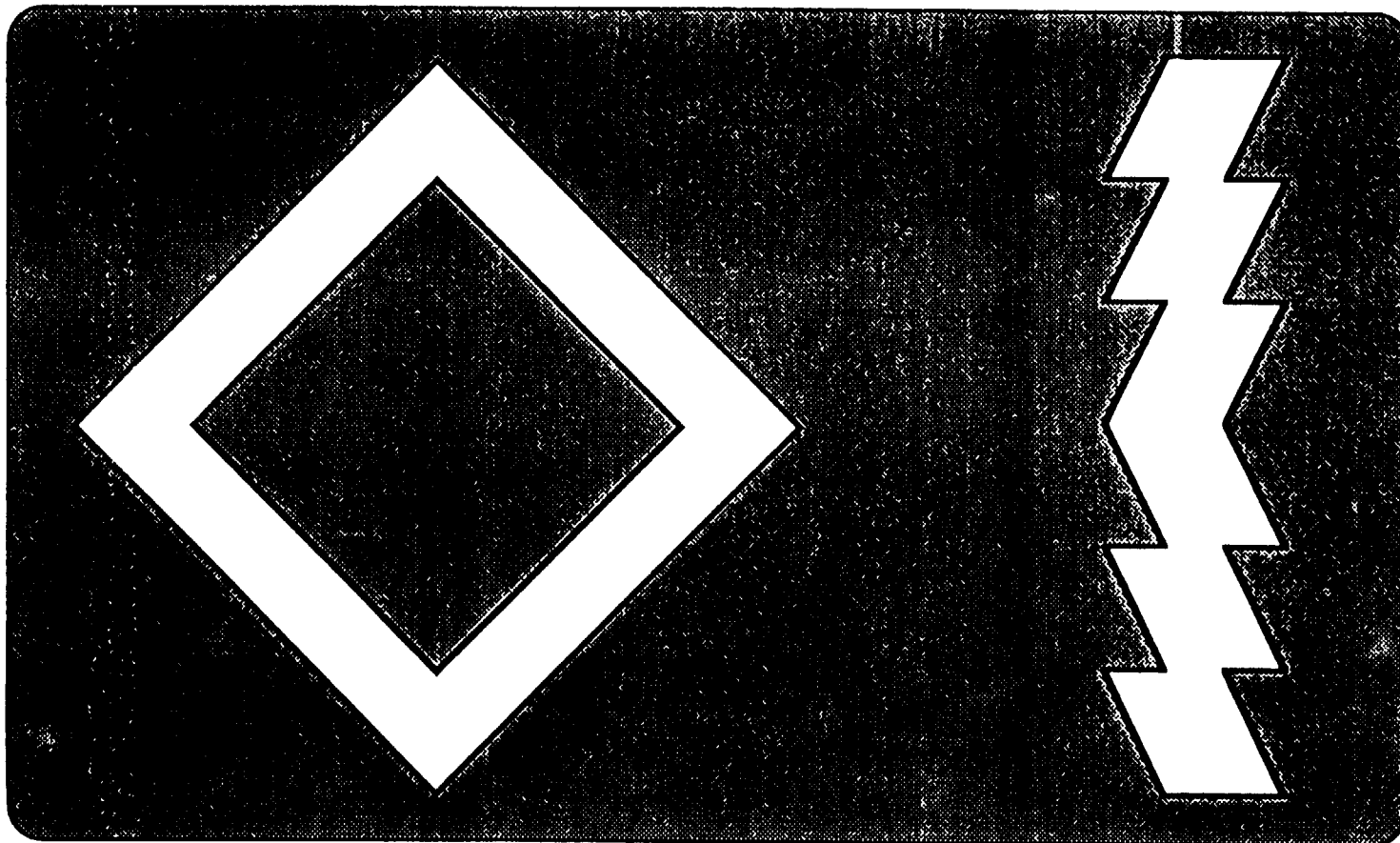
SHAPED RADIANCES YIELD HIGH SNRs FOR CALIBRATION VIS/NIR/SWIR



SPATIAL BAND-TO-BAND REGISTRATION RETICLE PATTERN CONCEPTS



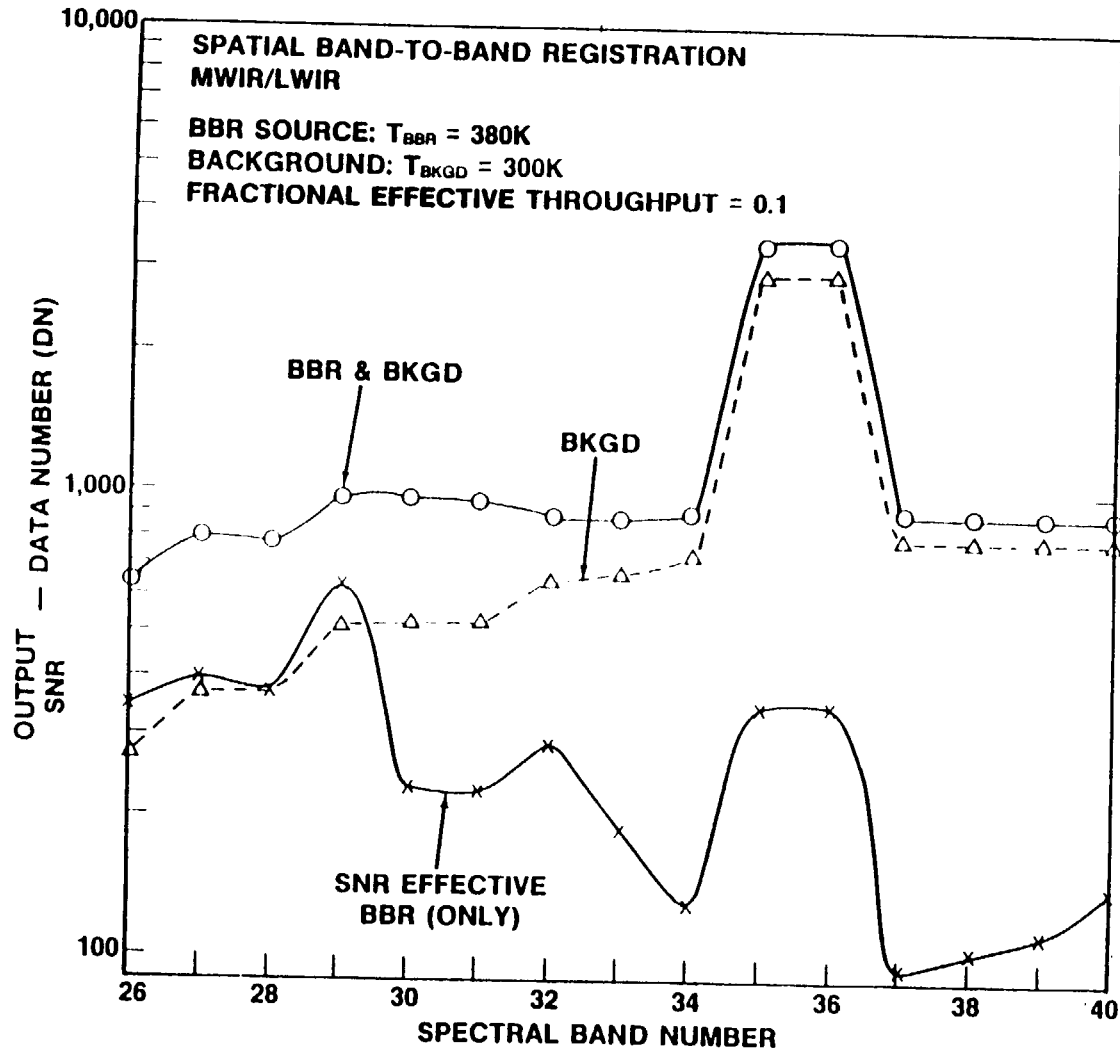
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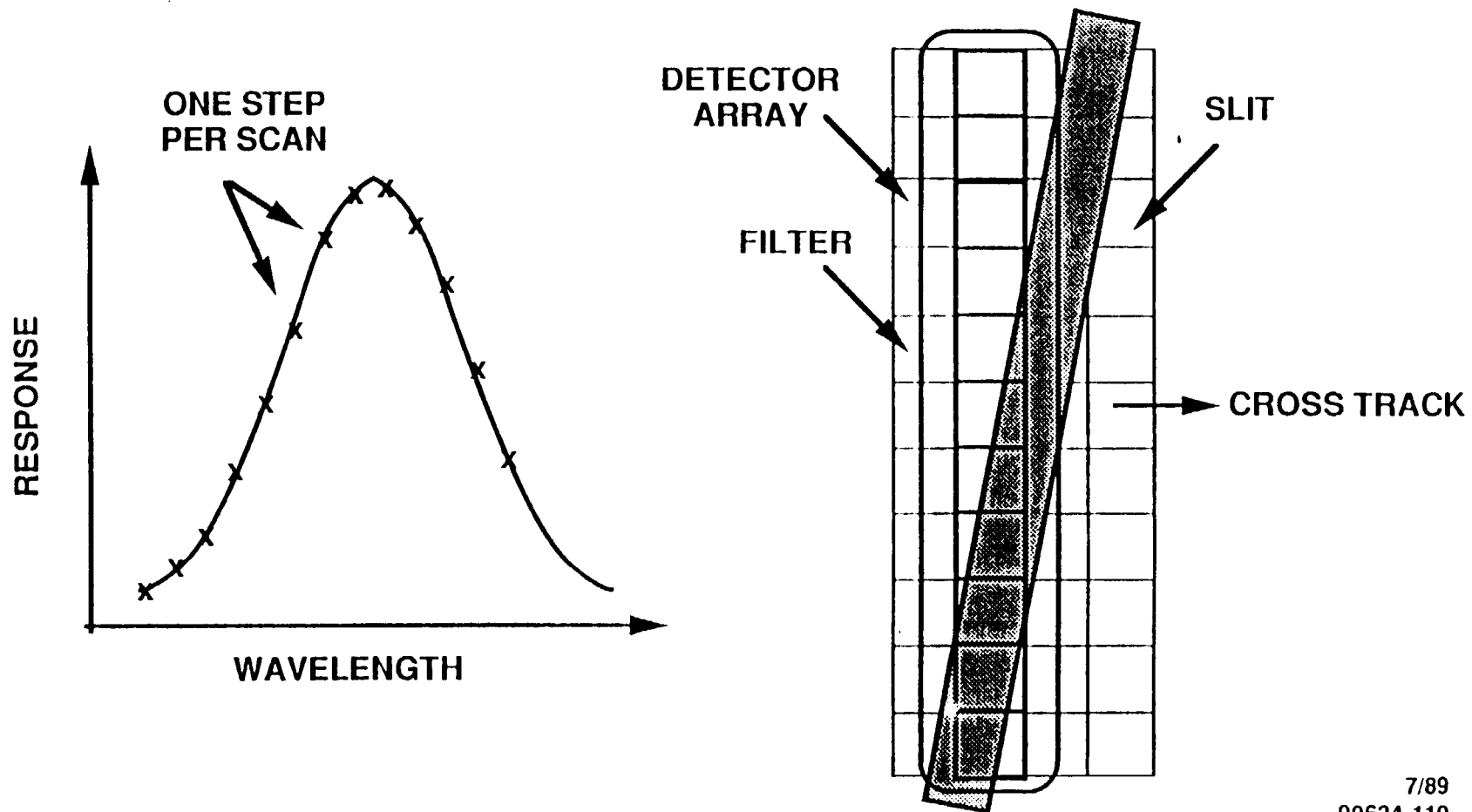


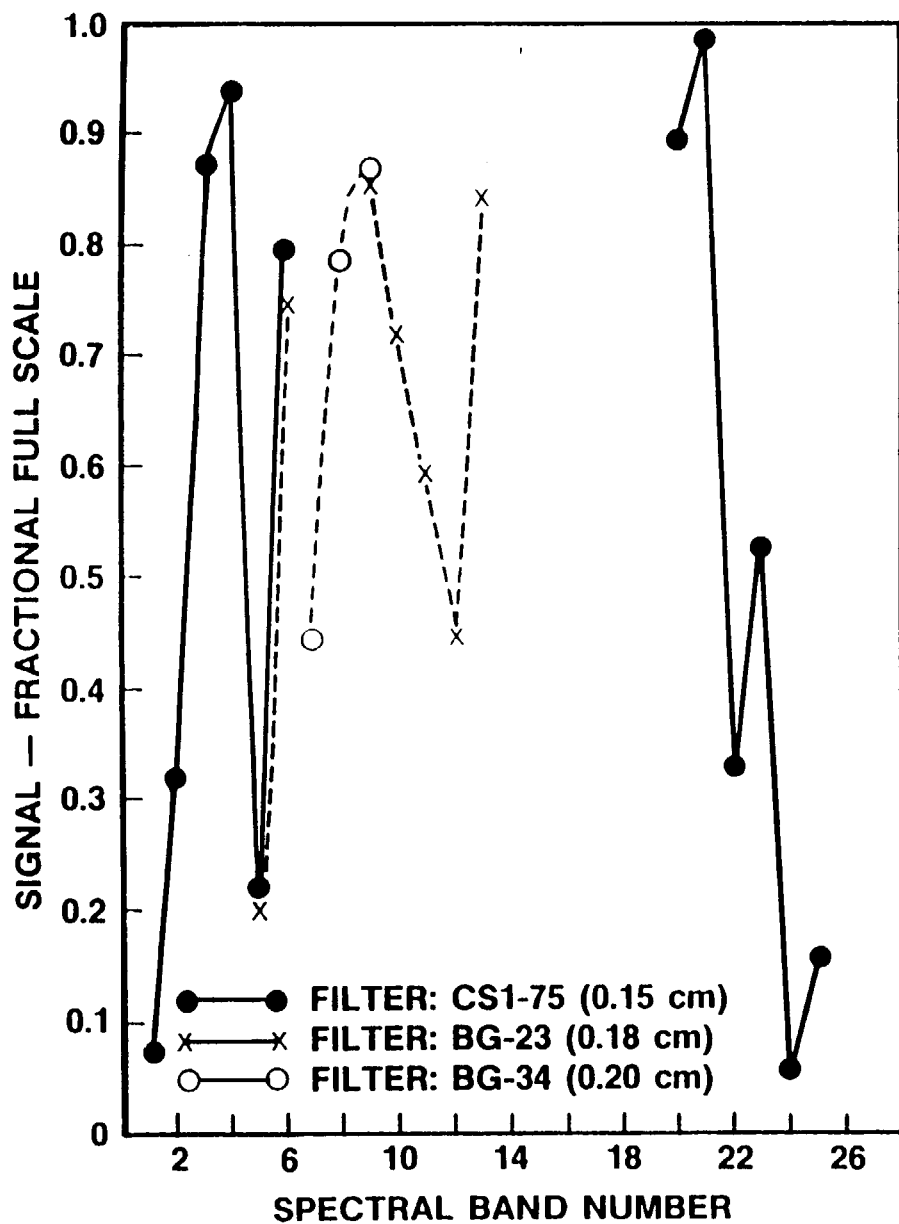
MWIR/LWIR 0.1 PIXEL ON-BOARD REGISTRATION MEASUREMENT ACHIEVABLE

RELATIVE SPECTRAL RESPONSE MEASUREMENT CONCEPT

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**FILTERS SHAPE
SRCA RADIANCE
FOR ON-BOARD
SPECTRAL
CALIBRATION**

MODIS-N FAR FIELD RESPONSE MEASUREMENT



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- CALIBRATION SOURCE DOES NOT SIMULATE EARTH (TYPICALLY)
- MODIS-N OPENING (AT APERTURE DOOR) WILL BE DIVIDED INTO THREE SECTIONS
- EARTH SIMULATION IS ATTAINABLE WITH SIS (48)

SUMMARY OF FAR FIELD SPURIOUS RESPONSE CONFIGURATIONS

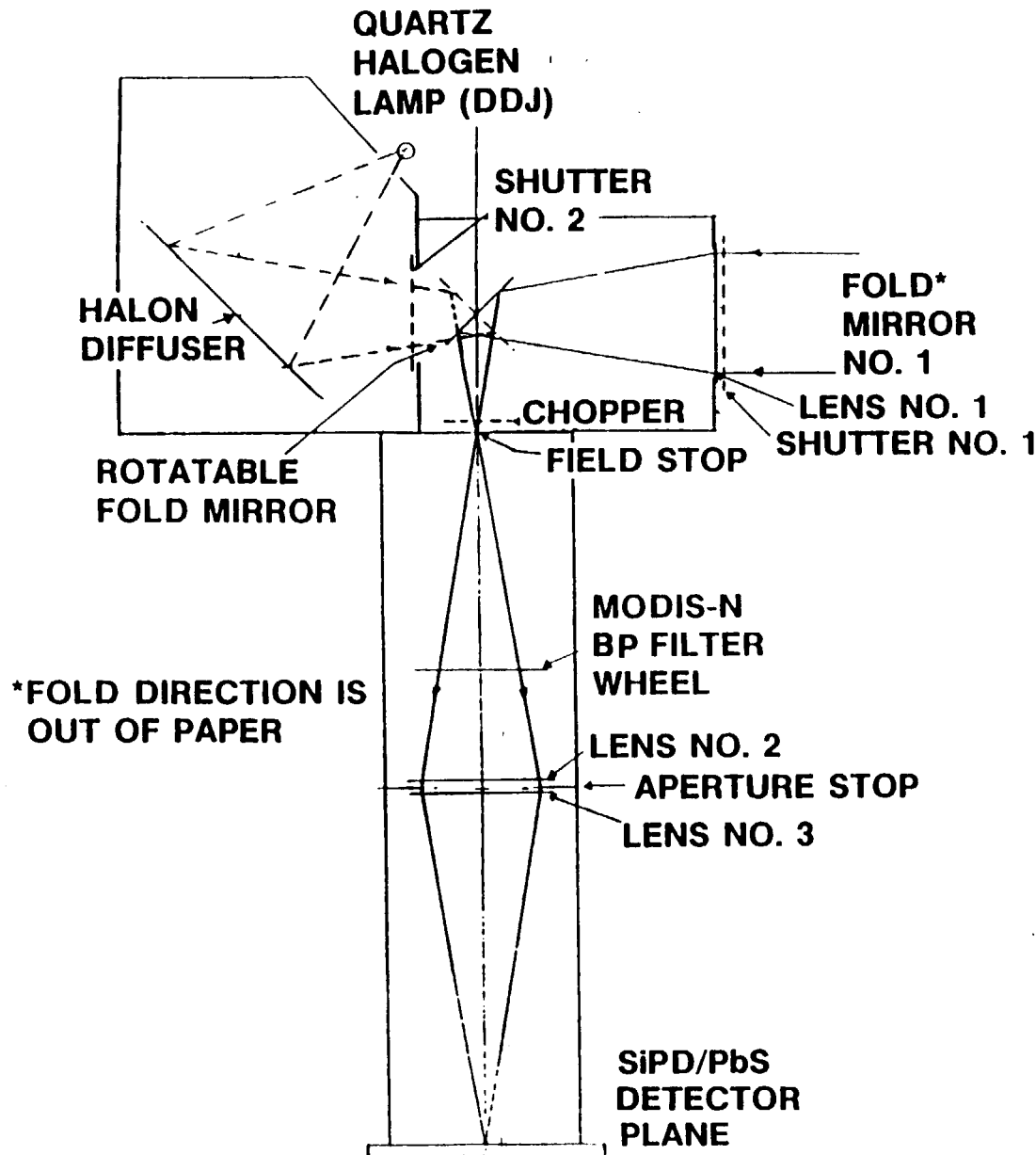
| SOURCE | CONFIGURATION | DISTANCE SOURCE TO MODIS-N | EFFECTIVE SOLID ANGLE (SR) |
|----------|------------------------|----------------------------|---|
| EARTH | ON ORBIT | 705 km | $\ \sin^2 (64.2) = 2.55$ |
| SIS (48) | FAR CONDITION | 960 INCHES | $\ \sin^2 [\text{Atn}(9/960)] = 0.00028$ |
| SIS (48) | NEAR CONDITION | 8 INCHES | $\ \sin^2 [\text{Atn}(9/8)] = 1.75$ |
| SIS (48) | NEAR (IDEAL) CONDITION | 4.34 INCHES | $\ \sin^2 [\text{Atn}(9/4.34)] = 2.55$ |

- POTENTIAL SCATTERED (REENTRANT) ENERGY CAN CONTAMINATE MEASUREMENT
- SIS (48) RADIANCE MONITOR



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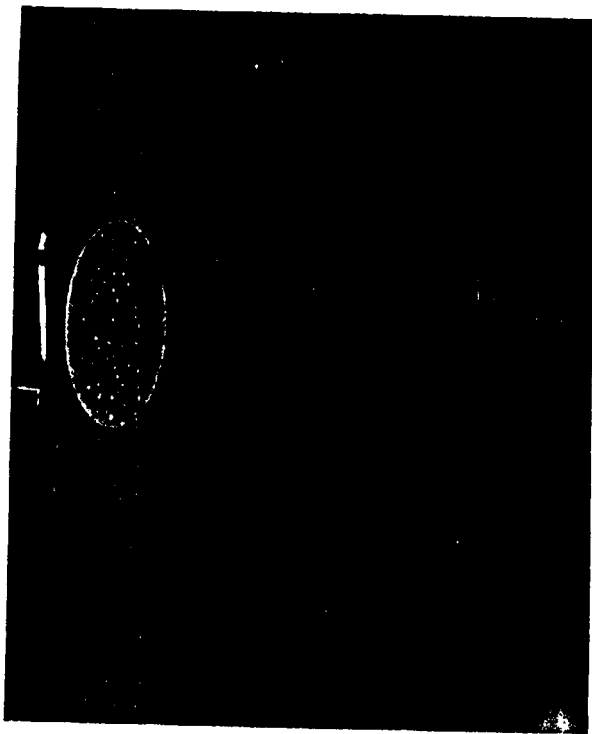
OPTICAL SCHEMATIC OF SIS(48) MONITORING SYSTEM



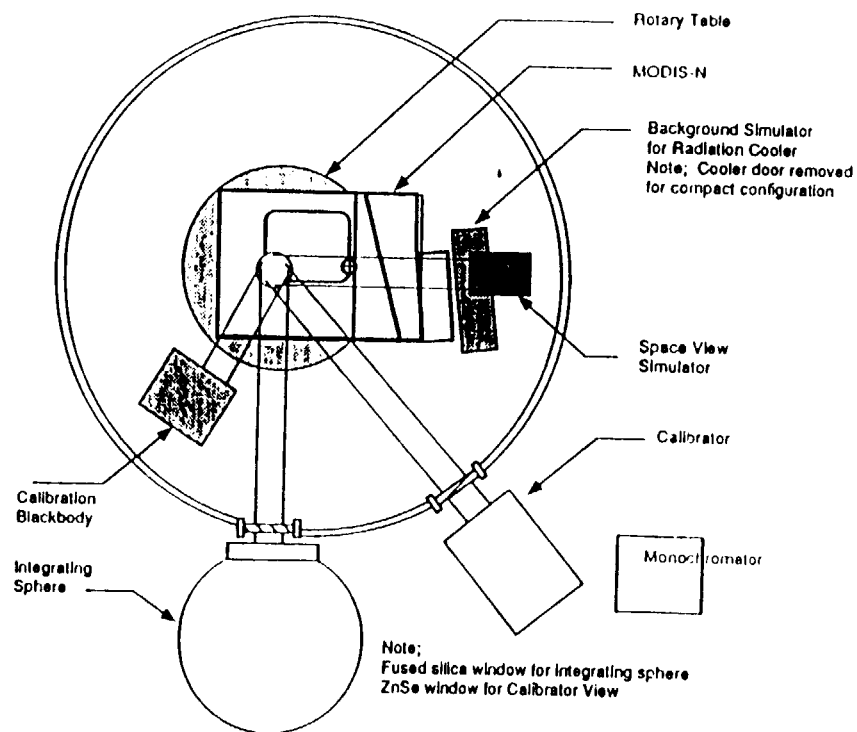
MODIS-N PRE-FLIGHT CALIBRATION HARDWARE AND TEST EXPERIENCE BASED ON TM

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- INTEGRATING SPHERE FOR VIS/NIR/SWIR RADIOMETRIC CALIBRATION



- THERMAL VACUUM CHAMBER TEST CONFIGURATION

MODIS-N CALIBRATION UNCERTAINTY ESTIMATE



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SIS RADIOMETRIC CALIBRATION UNCERTAINTY

| | |
|-------------------------------|------|
| Irradiance/radiance standards | 0.5% |
| Transfer to working standard | 1.0% |
| Halon diffuser | 0.5% |
| Monochromator comparator | |
| Scattered light | 1.0% |
| Wavelength calibration | 0.3% |
| Nonuniform response | 1.0% |
| Geometric factors | 1.0% |
| SIS | |
| Current control | 0.5% |
| Nonuniformity | 0.5% |
| SNR | |
| Standard lamp usage | |
| Orientation | 0.3% |
| Diffuser to lamp | 0.5% |
| Current | 0.5% |
| Linearity | 1.5% |
| RSS total | 2.8% |

MODIS-N REFLECTANCE BANDS CALIBRATION ACCURACY ESTIMATE

| | |
|------------------------------|------|
| SIS calibration | 2.8% |
| Chamber window | 1.0% |
| SNR | 0.5% |
| Nonuniformity | 0.5% |
| Stray light after correction | 1.0% |
| Polarization | 0.5% |
| Spectral effects | |
| In-band | 0.5% |
| Out of band | 1.0% |
| RSS total | 3.4% |
| Margin | 2.0% |
| Total | 3.9% |
| Requirement | 5.0% |

SPECTRAL BAND SHIFTS ARE A MAJOR CONCERN

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- **PAST INSTRUMENTS**
 - MSS BAND 3 LONG WAVELENGTH EDGE SHIFTED 9 nm
 - MSS BANDS HAD RADIOMETRIC CALIBRATION SHIFT FROM -10 TO 12% AIR TO VACUUM
 - SPOT HAS HAD EFFECTS IN 3 TO 27% DOMAIN
 - SPECTRAL PROPERTIES OF BAND PASS FILTERS HAVE BEEN SENSITIVE TO ABSORBED WATER
 - ABSORBED/DESORBED WATER EFFECTS ARE REVERSIBLE WITH VARIABLE TIME CONSTANTS

SPECTRAL BAND SHIFTS ARE A MAJOR CONCERN (CONT)

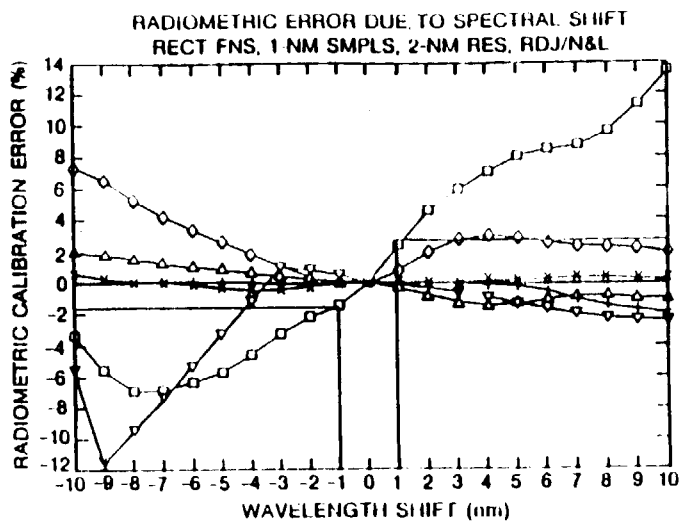
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- **SOLAR ILLUMINATED DIFFUSER (OBC)**
 - SPECTRAL VARIATIONS OF SOLAR IRRADIANCE CAUSED BY FRAUNHOFER LINES
 - VARIATIONS ARE PREDOMINANTLY IN THE REGION OF 0.4 TO 0.68 μm
 - BAND-DEPENDENT WITH VARIATION <0.5% TO 4% FOR 2 nm SHIFT
- **EARTH/ATMOSPHERIC SCENES**
 - BAND 12 (765 nm) HAS 25% VARIATION OF MODELED REFLECTANCE FOR A 5 nm SHIFT
 - SOME BANDS ARE INSENSITIVE TO SHIFTS

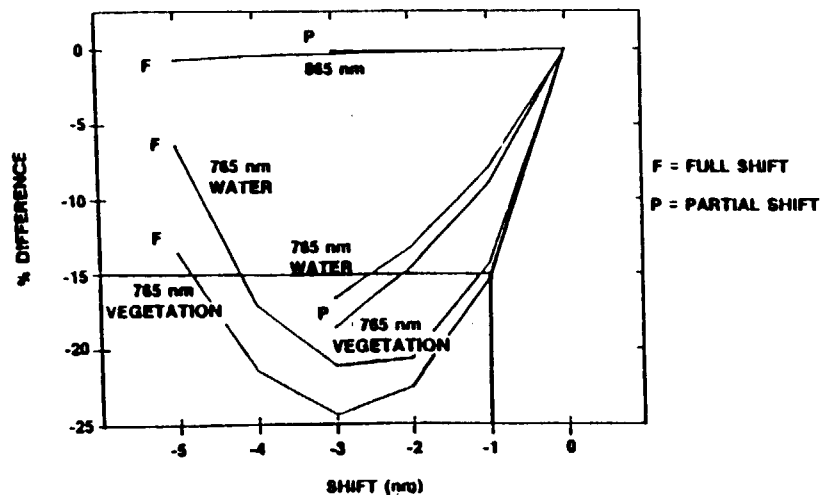
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◇ 10 AT 490 nm (BAND 6) △ 10 AT 520 nm (BAND 7) × 10 AT 565 nm (BAND 8)
+ 20 AT 470 nm (BAND 1) □ 10 AT 435 nm (BAND 5) ▽ 10 AT 470 nm

RADIOMETRIC ERROR ASSOCIATED WITH SOLAR SPECTRAL STRUCTURE



RADIOMETRIC ERROR ASSOCIATED WITH GROUND SPECTRAL SIGNATURES

**RADIOMETRIC ERROR
ASSOCIATED WITH
SPECTRAL
SIGNATURES**

MODIS-N SPECTRAL SHIFTS AND THEIR EFFECTS WILL BE MINIMIZED

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- o MINIMIZING THE EFFECTS OF SPECTRAL SHIFTS WILL BE DONE VIA A MULTIPLE APPROACH
- o STATE OF THE ART OPTICAL FILTERS WILL BE MANUFACTURED TO MINIMIZE SPECTRAL INSTABILITY : $\Delta\lambda < 2$ nm
- o GBC SRCA WILL BE USED TO CHARACTERIZE MODIS-N SPECTRAL BANDS ($\lambda < 1.0$ μm) SHIFTS ON ORBIT
- o SPECTRAL BANDS ($\lambda > 1.0$ μm) WILL NOT BE CHARACTERIZED ON ORBIT
 - o Optical filters sealed in vacuum detector dewar
 - o Filters not subjected to air-vacuum cycling
 - o Detector - filter combination is well temperature controlled
- o SOLAR ILLUMINATED DIFFUSER WILL BE USED TO CALIBRATE MODIS-N REFLECTANCE BANDS ON ORBIT

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OBC — DIFFUSER SELECTION CRITERIA

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- BRDF:** 0.18 sr⁻¹ AND 0.018 sr⁻¹
- STABILITY:** EFFECTS OF RADIATION BELTS
EFFECTS OF SOLAR ULTRAVIOLET IRRADIATION
EFFECTS OF CONTAMINATION
- DURABILITY:** LAUNCH VIBRATION AND SHOCK LOADS
- POLARIZATION:** DEGREE OF POLARIZATION FOR REFLECTED
COMPONENT
- CANDIDATE MATERIALS:** HALON WITH CARBON FILLER
CROSSED CYLINDRICAL LENS REFLECTOR ARRAYS
WITH EVAPORATED NDF LAYERS
GROUND METAL SURFACES WITH EVAPORATED THIN
FILMS OVERCOAT
YB 71 WHITE PAINT
- PRELIMINARY SELECTION:** YB 71 WHITE PAINT

LUNAR VIEW AVAILABLE THROUGH SPACE-VIEW PORT

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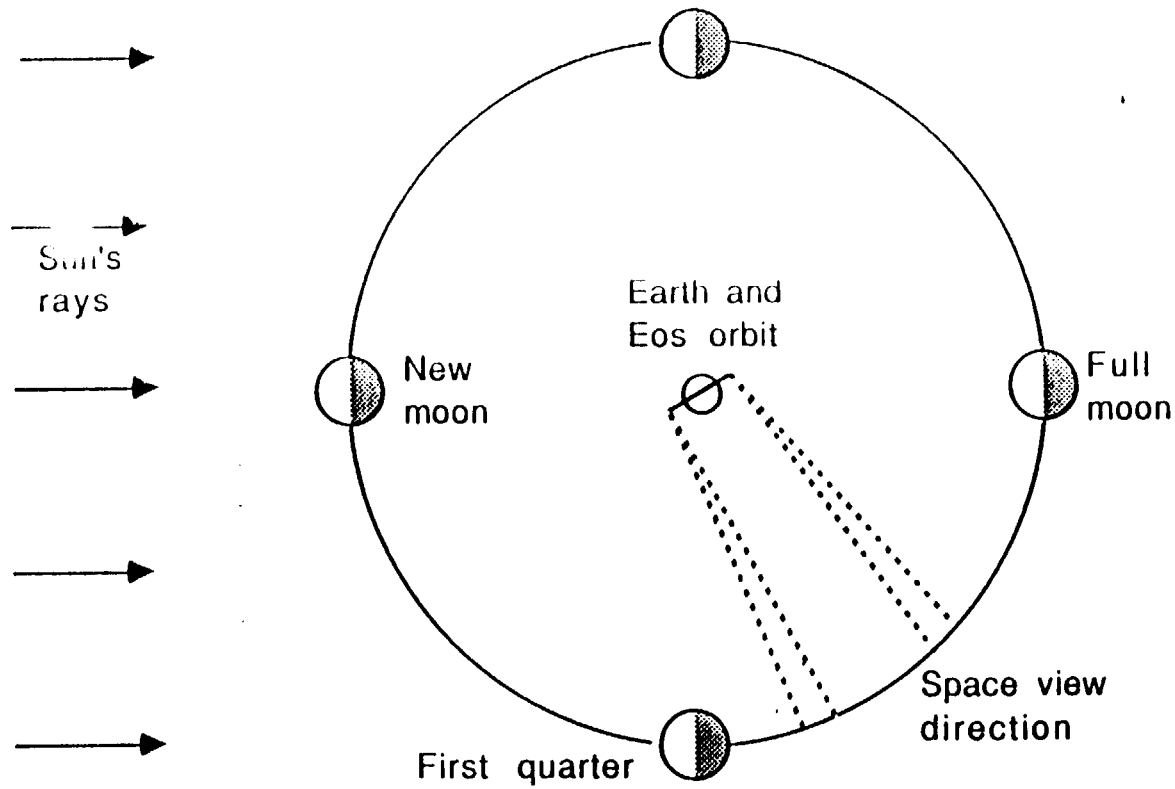
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- THE MOON IS VISIBLE AT PREDICTABLE TIMES THROUGH THE MODIS-N DC-RESTORE SPACE-VIEW PORT
- SIMPLE ELECTRONIC GATING WILL PREVENT LUNAR INTERFERENCE WITH DC RESTORE FUNCTION AND ALLOW ACQUISITION OF LUNAR RADIANCE DATA FOR CALIBRATION STABILITY CHECKS
- LUNAR VIEWING CONDITIONS:
 - MOON PHASE WILL BE BETWEEN FIRST QUARTER AND FULL
 - MOON VISIBLE IN SPACE-VIEW PORT 4 - 6 TIMES PER YEAR, NOT AT EVEN INTERVALS
 - WHEN IN VIEW, MOON VISIBLE APPROXIMATELY A DAY FOR PART OF EACH ORBIT
 - MOON SAMPLED 5 TO 20 TIMES EACH SCAN

GEOMETRY FOR LUNAR VIEW THROUGH SPACE PORT

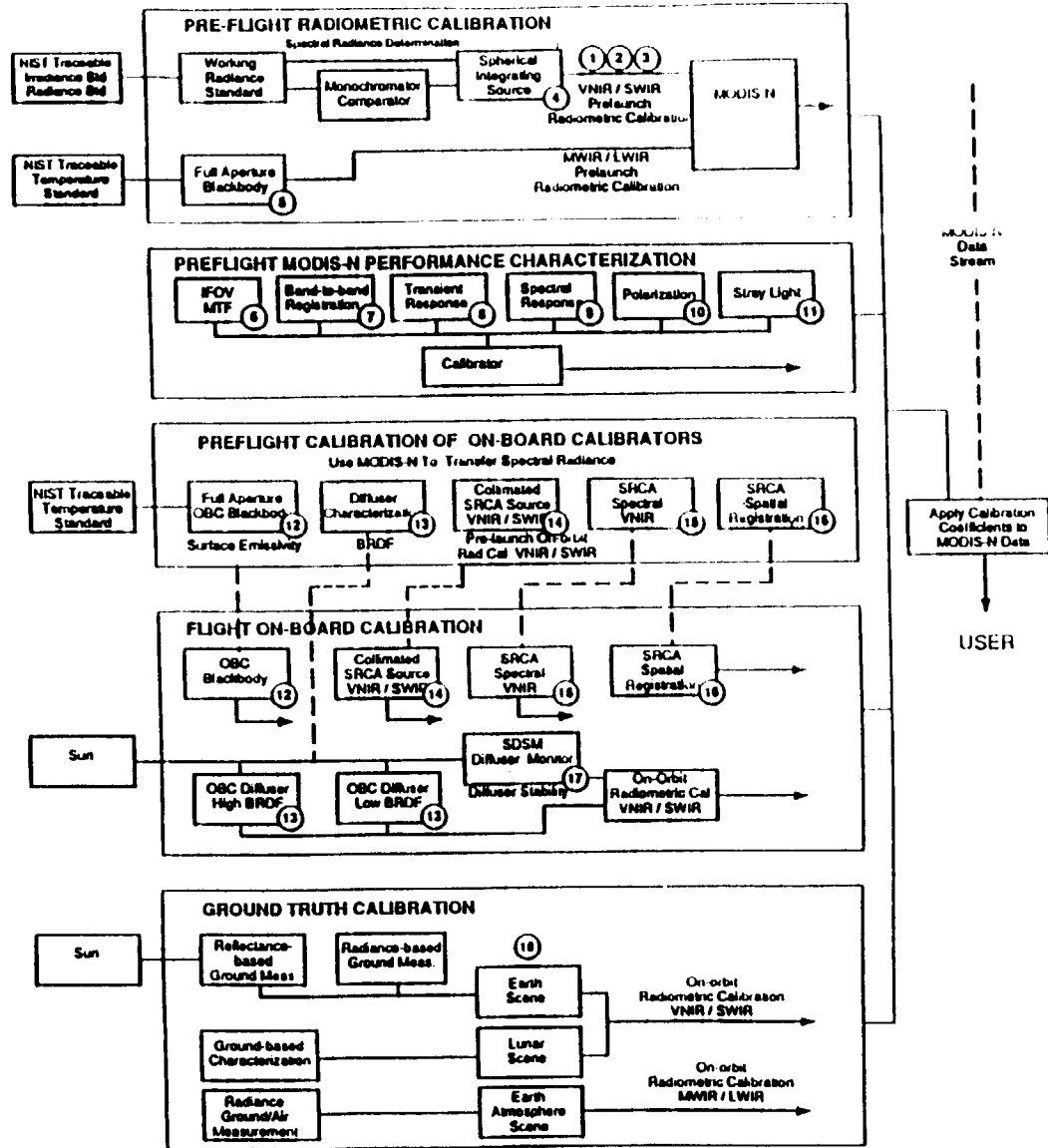
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MODIS-N CALIBRATION MANAGEMENT PLAN

MODIS-N CALIBRATION MANAGEMENT PLAN BLOCK DIAGRAM COVERING
PREFLIGHT AND ON-ORBIT ACTIVITIES INCLUDING MATH MODEL INDEX



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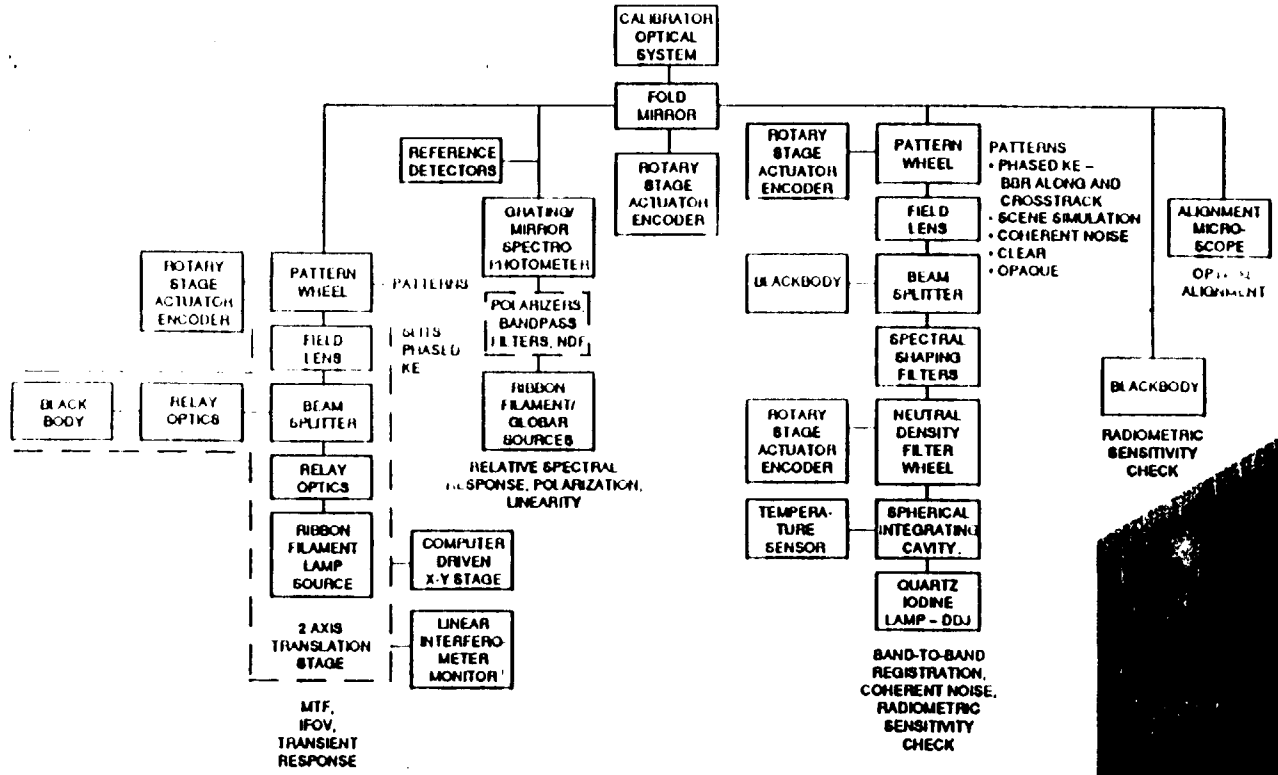
PERFORMANCE CHARACTERIZATION AND CALIBRATION MATH MODEL

| Model | Purpose | Model Number |
|--|---|--------------------------|
| Transfer spectral radiance/irradiance to spherical integrator source (SIS) via monochromator comparator – multiple methodologies | Source calibration, VIS/NIR/SWIR | 1a, 2a, 3a 1b, 2b, 3b |
| SIS – MODIS-N | Instrument calibration, VIS/NIR/SWIR | 4a, 4b |
| Full aperture blackbody – MODIS-N | Instrument calibration, MWIR/LWIR | 5 |
| MTF/IFOV – calibrator (phased reticles) – MODIS-N | Instrument characterization, all bands | 6 |
| Band-to-band registration – calibrator and phased reticle (CT/AT) – MODIS-N | Instrument characterization, all bands | 7 |
| Transient response – calibrator – MODIS-N | Instrument characterization, all bands | 8 |
| Spectral response – monochromator – MODIS-N | Instrument characterization, all bands | 9 |
| Polarization – calibrator – MODIS-N | Instrument characterization, all bands | 10 |
| Stray light response/calibrator/SIS/large aperture IR source – MODIS-N | Instrument characterization, all bands | 11a, 11b |
| OBC blackbody – MODIS-N | Instrument calibration, MWIR/LWIR | 12 |
| OBC diffuser calibration | BRDF 0.4-2.40 μm , angle | 13a |
| OBC solar-illuminated diffuser – MODIS-N | Instrument calibration, VIS/NIR/SWIR | 13b |
| SRCA – filtered incandescent source – MODIS-N | Calibration transfer, VIS/NIR/SWIR | 14a |
| SRCA – filtered incandescent source – MODIS-N | Instrument calibration check, VIS/NIR/SWIR | 14b |
| Wavelength calibration – SRCA monochromator | λ calibration, VIS/NIR | 15a |
| Spectral response – SRCA – MODIS-N | Instrument characterization, VIS/NIR | 15b |
| Band-to-band spatial registration – OBC SRCA – MODIS-N | Instrument characterization, all bands | 16 |
| Diffuser – sun – solar illuminated diffuser stability monitor | Diffuser stability check, 0.43-0.90 μm | 17 |
| Ground truth – Earth/lunar/atmosphere – MODIS-N – several major models | Instrument calibration | 18 |

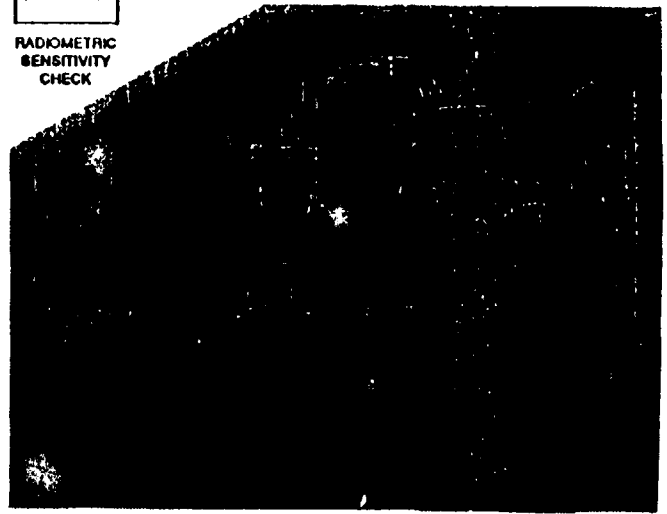
MODIS-N CALIBRATOR WILL BE USED TO CONDUCT A BATTERY OF TESTS



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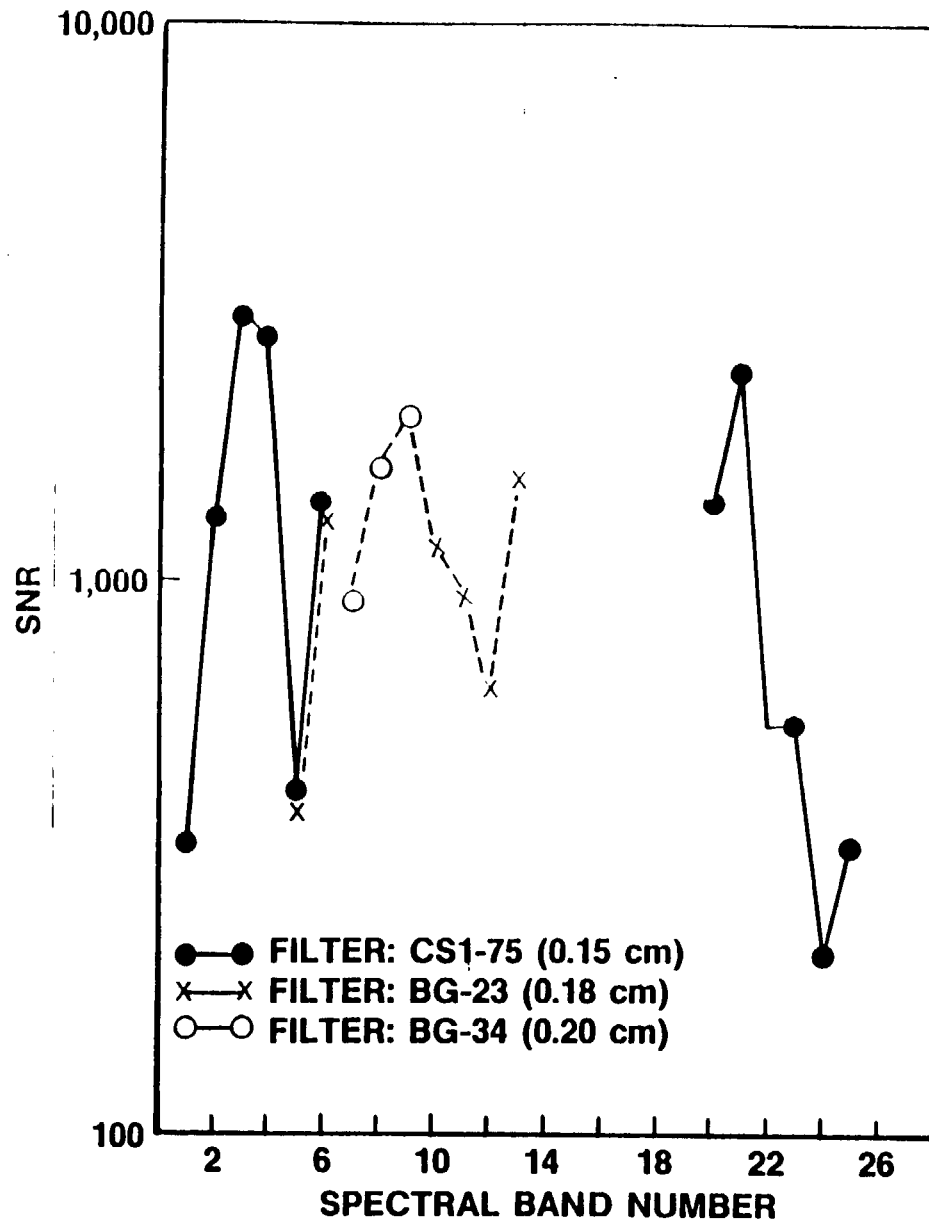


TM CALIBRATOR





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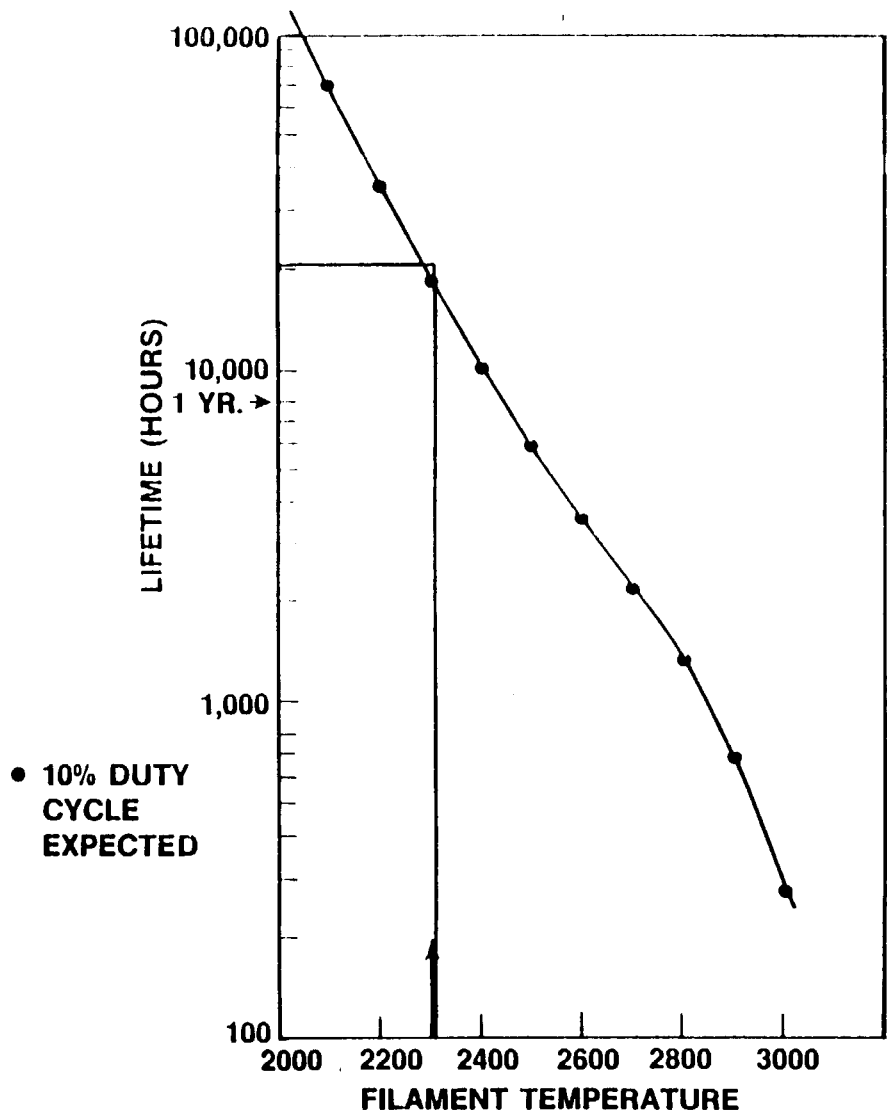


SHAPED RADIANCES YIELD HIGH SNRS FOR SPECTRAL CALIBRATION

6/89
90634-93



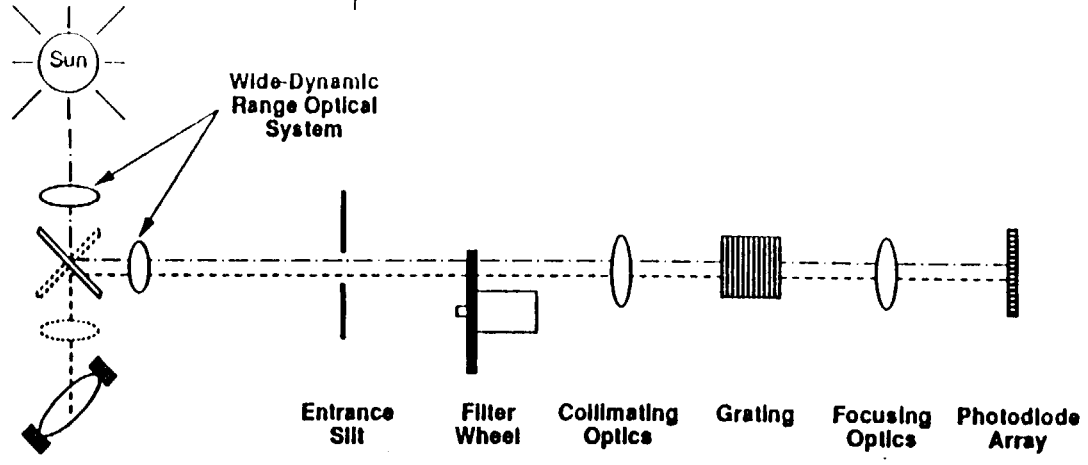
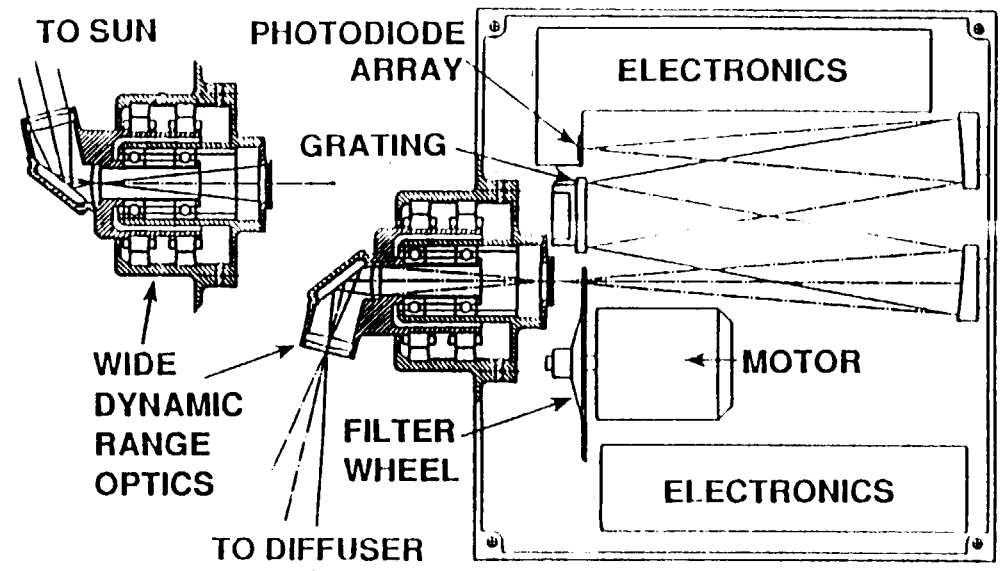
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INCANDESCENT SOURCE EXPECTED TO HAVE LIFETIME GREATER THAN 10,000 HOURS



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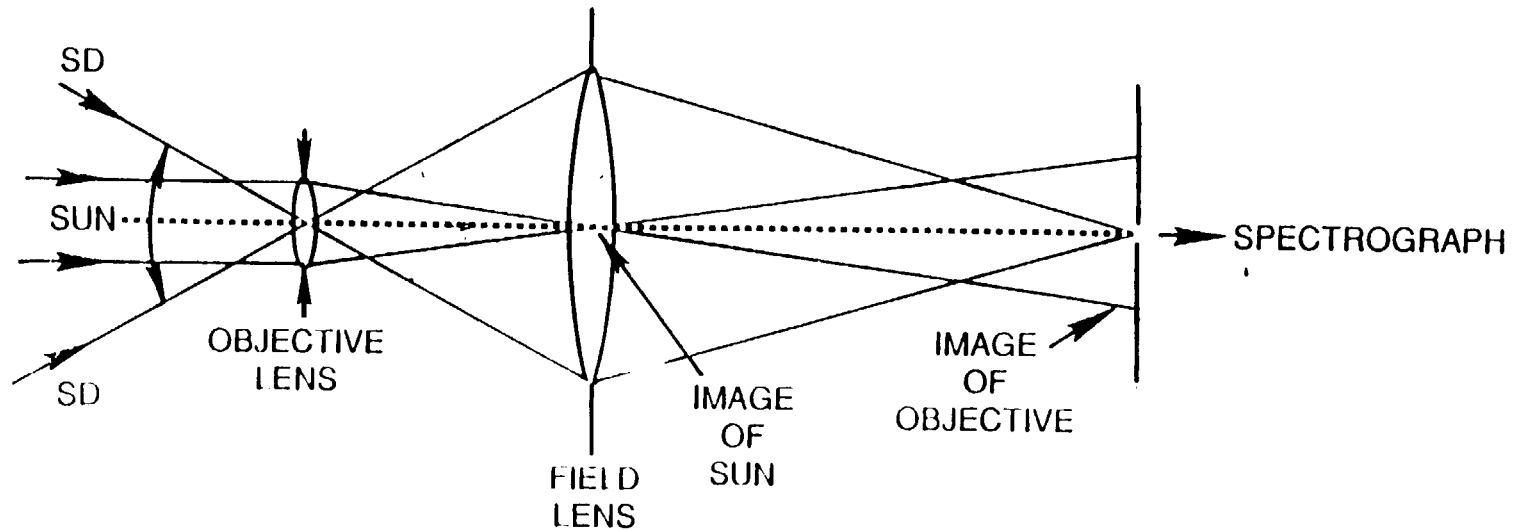


Solar Diffuser

• Solar Diffuser Reflectance Monitored
Using Sun as Reference

SOLAR DIFFUSER STABILITY MONITOR (SDSM) ALTERNATELY VIEWS SUN AND DIFFUSER

WIDE DYNAMIC RANGE OPTICAL SYSTEM

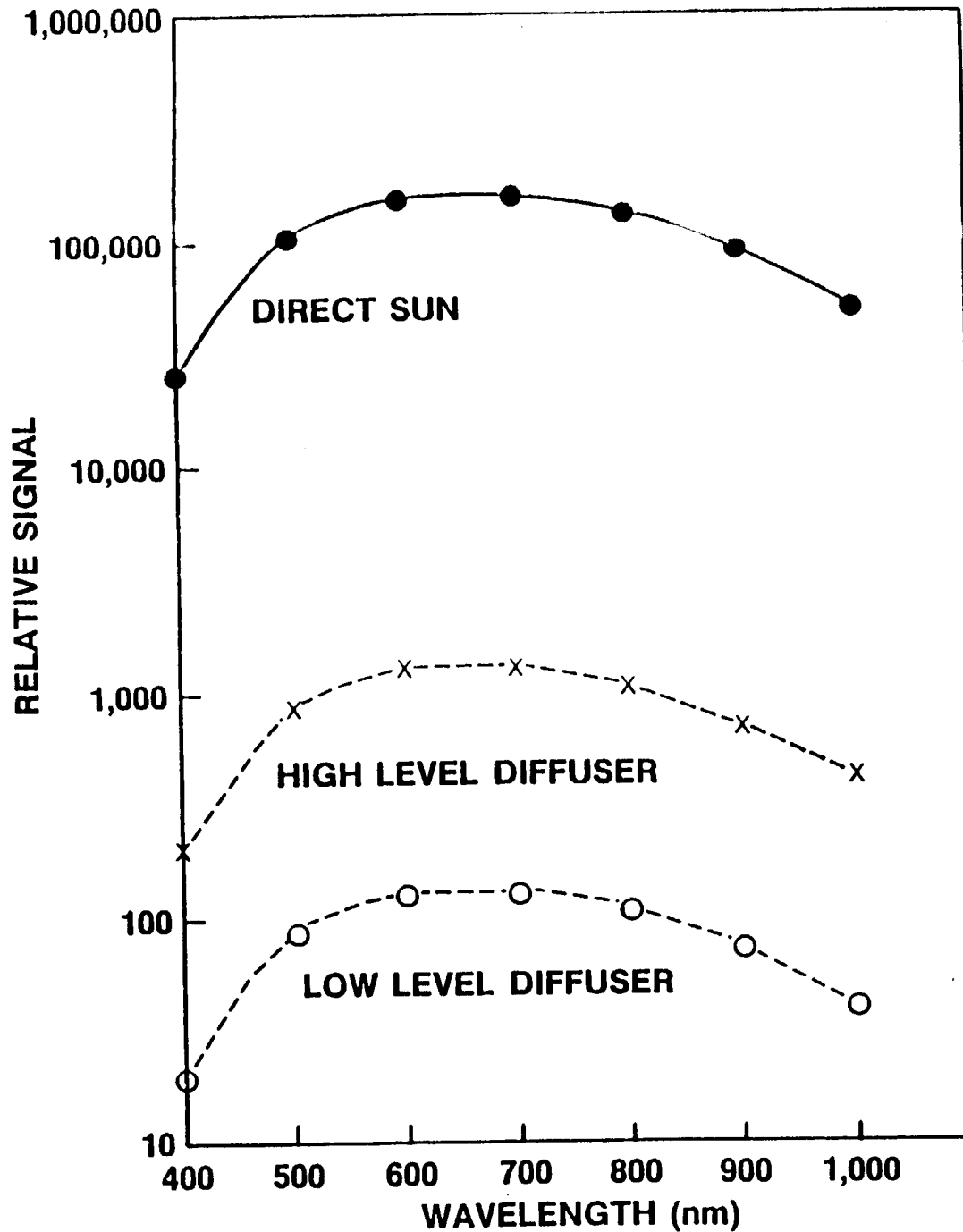


$$\frac{P_{\text{SUN}}}{P_{\text{SD}}} = \frac{4 (f/\#)_{\text{SD}}^2}{\pi \cdot \text{BRDF} \cdot \text{COS}(\phi_i)} \approx 90$$

- VARIABLE VIGNETTING DUE TO DIFFRACTION: $< 1/2\%$
- NONUNIFORM SENSITIVITY OF SPECTRORADIOMETER: CHARACTERIZE
- VARIABLE ILLUMINATION ANGLES: CHARACTERIZE



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**SOLAR ILLUMINATED
DIFFUSER MONITOR
RELATIVE SIGNAL
AS A
FUNCTION OF
MEASUREMENT MODE**

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| Type of Calibration | Source | Mechanism | Aperture | Spectral Bands | Usage Frequency (Max) | Other Comments |
|----------------------------|-----------------------------------|---|----------|---|---------------------------------|--|
| DC restore | Space | | Full | All | Once per scan line | |
| Radiometric | Sun | Solar illuminated diffuser | Full | VIS/NIR/SWIR less 8 through 16 | Once per orbit | BRDF = 0.18 sr ⁻¹ |
| Radiometric | Sun | Solar illuminated diffuser | Full | VIS/NIR/SWIR | Once per orbit | BRDF = 0.18 sr ⁻¹ |
| Radiometric | Blackbody | Blackbody | Full | MWIR/LWIR | Once per scan line | |
| Radiometric | Incandescent source | SRCA spectrally shaped collimator | Partial | VIS/NIR/SWIR | Available any time during orbit | |
| Spatial Registration | Incandescent source and IR source | SRCA | Partial | VIS/NIR/SWIR MWIR/LWIR | Available any time during orbit | |
| Spectral (MODIS-N) | Incandescent source | SRCA grating monochromator | Partial | VIS/NIR | Available any time during orbit | Grating is rotated to produce λ scan |
| Spectral (monochromator) | Incandescent source | SRCA grating monochromator and filter | Full | $0.40 \mu\text{m} \leq \lambda \leq 1.00 \mu\text{m}$ | Available any time during orbit | Grating is rotated to produce λ scan |
| Diffuser stability monitor | Sun | SDSM grating spectrograph and fold mirror | Full | $0.40 \mu\text{m} \leq \lambda \leq 0.90 \mu\text{m}$ | Available once per orbit | High BRDF diffuser |
| Diffuser stability monitor | Sun | SDSM grating spectrograph and fold mirror | Full | $0.40 \mu\text{m} \leq \lambda \leq 0.90 \mu\text{m}$ | Available once per orbit | Low BRDF diffuser |

MODIS-N IN-FLIGHT CALIBRATION CAPABILITY

GSE/BTE FUNCTIONS

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- SIMULATE INTERFACE BETWEEN MODIS-N AND HOST SPACECRAFT
- PROVIDE OPTICAL STIMULI FOR PERFORMANCE TESTING AND CALIBRATION
- PROCESS HIGH RATE DATA STREAM
- MONITOR, CONTROL AND RECORD TEST CONFIGURATIONS AND OPERATIONAL CONDITIONS
- PROVIDE COOLING OF SWIR, MWIR, AND LWIR FOCAL PLANES FOR LABORATORY AMBIENT TESTS
- PROVIDE RADIATIVE HEAT SINKS FOR THERMAL VACUUM TESTS
- PROVIDE SPECIAL TOOLS AND FIXTURES FOR ALIGNMENT AND HANDLING

FUNCTIONAL/
PERFORMANCE PARAMETER

| | Ambient | Vacuum | MCCIS-N Calibrator | Reticles | Slit source | Grating Monochromator | Payload Stimulus Control Console | Spherical Integrator Source | Blackbody FAIRS | Large-Aperture IR Source LAIRS | DC Resonance Source | Alignment Test Set | Bench Cooler | Space Background Simulator | Payload Interface Control Console | Computer | High Data Rate Recorder | TEST PHASE* |
|----------------------------|---------|--------|--------------------|----------|-------------|-----------------------|----------------------------------|-----------------------------|-----------------|--------------------------------|---------------------|--------------------|--------------|----------------------------|-----------------------------------|----------|-------------------------|------------------|
| Command and Power | | | | | | | | | | | | | | | | | | |
| Command Verification | X | X | | | | | | | | | | | | | X | X | | 1, 2, 4, 5, 6 |
| Power Profile | X | X | | | | | | | | | | | | | X | X | | |
| Turn On Transients | X | X | | | | | | | | | | | | | X | X | | |
| Operational Transients | X | X | | | | | | | | | | | | | X | X | | |
| Power Supply Transients | X | X | | | | | | | | | | | | | X | X | | |
| Power Supply Ripple | X | X | | | | | | | | | | | | | X | X | | |
| Spectral Coverage | | | | | | | | | | | | | | | | | | |
| In-Band Shape | X | X | X | | | X | | | | | X | | X | X | X | X | | 1, 2, 3, 4, 5 |
| Out of Band | X | X | X | | | X | | | | | X | | X | X | X | X | | |
| Spatial Coverage (IFOV) | | | | | | | | | | | | | | | | | | |
| Cross Track | X | | X | X | X | | | | | | X | | | X | X | X | | 1, 2, 3 |
| Along Track | X | | X | X | X | | | | | | X | | | X | X | X | | |
| Far Field Spurious | X | | X | | | | | X | X | X | X | | X | | X | X | | 4 |
| Radiometric Performance | | | | | | | | | | | | | | | | | | |
| Gain | X | X | X | X | | | X | X | X | X | X | X | X | X | X | X | X | 3, 4, 5, 6 |
| Offset | X | X | X | X | | | X | X | X | X | X | X | X | X | X | X | X | |
| Signal Versus Radiance | X | X | X | X | | | X | X | X | X | X | X | X | X | X | X | X | |
| Linearity | X | X | X | X | | | X | X | X | X | X | X | X | X | X | X | X | |
| SNR | X | X | X | X | | | X | X | X | X | X | X | X | X | X | X | X | |
| On Board Calibrator | | | X | X | X | | X | X | X | | X | | X | X | X | X | X | |
| Spectral Matching | | | X | X | X | | | | | | | | | | | | X | |
| Coherent Noise | X | X | X | X | | | X | X | X | | X | | X | X | X | X | X | |
| Scan Modulation | X | X | | | | | | | | | X | | X | X | X | X | X | |
| Band-to-Band Stability | X | X | | X | | | | X | X | | X | | X | X | X | | X | |
| MTF (Sine Wave) | | | | | | | | | | | | | | | | | | |
| Along Track | X | X | X | X | | | X | | | | X | | X | X | X | X | | 1, 2, 3, 4, 5, 6 |
| Cross Track | X | X | X | X | | | X | | | | X | | X | X | X | X | | |
| Transient Response | | | | | | | | | | | | | | | | | | |
| Rise Time | X | X | X | X | | | X | | | | X | | X | | X | X | | 1, 4 |
| Overshoot/Undershoot | X | X | X | X | | | X | | | | X | | X | | X | X | | |
| Polarization Sensitivity | X | | | | | X | | | | | X | | X | | X | | | 1, 3 |
| Spectral Band Registration | | | | | | | | | | | | | | | | | | |
| Along Track | X | X | X | X | | | X | | | | X | | X | X | X | X | X | 3, 4, 5 |
| Cross-track | X | X | X | X | | | X | | | | X | | X | X | X | X | X | |
| Alignment Reference | X | | | | | | | | | | | X | | | | | | 3, 6 |

- * Test Phase
 1. Subsystem checkout
 2. Subsystem integration
 3. Optical alignment and performance (performed during subsystem integration and prior to Phase 4)
 4. Ambient environmental bench tests
 5. Thermal vacuum environmental tests
 6. Instrument/spacecraft integration tests



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PERFORMANCE VERIFICATION MATRIX

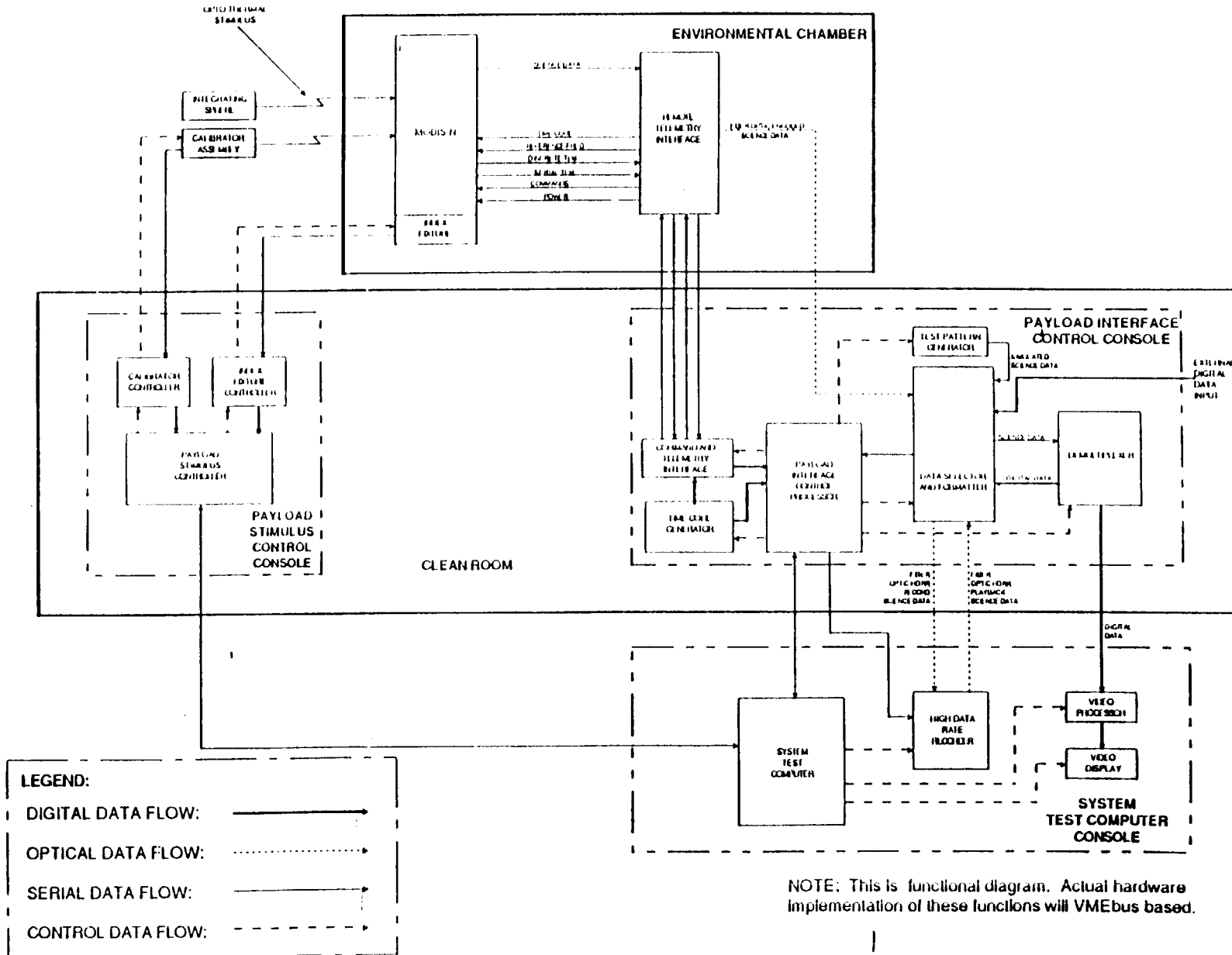


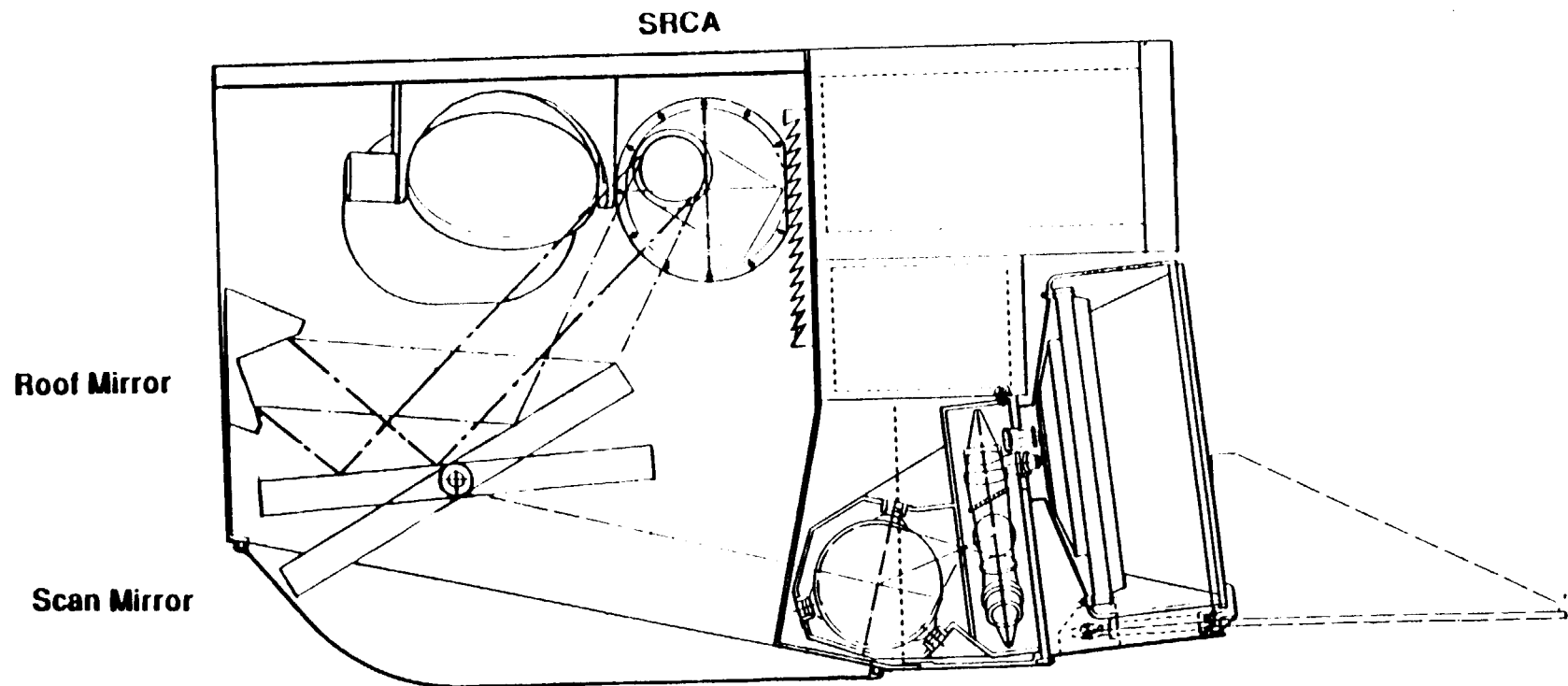
Figure 3-10. MODIS-N System Test Equipment Functional Diagram.

SCAN MIRROR REFLECTANCE CHARACTERIZATION

METHOD A - ROOF MIRROR

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CONCEPTUAL DESIGN FOR REFLECTANCE CHECK ACROSS THE SURFACE OF THE SCAN MIRROR

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- **Concern: Non-Uniform Contamination of the Scan Mirror On-Orbit**
- **Status: Baseline design for On-Orbit calibration sources does not provide information over the length of the Scan Mirror.**
- **Conceptual Solution: Characterize Spectral Reflectance of Scan Mirror across its length.**
- **Two Methods: Both use the SRCA Monochromator (VIS/NIR) with the following modifications:**
 - 1) **The collimator fold mirror must be capable of being tilted in small steps, in the scan direction, with a stepper motor.**
 - 2a) **A Reflecting Roof Mirror must be located within the scan cavity in a location near the end or the Earth Scan. The Scan Mirror is viewed at large angles of incidence.**
 - 3a) **A Silicon Photodiode must be added at the end of the SRCA exit slit to sense the Reflected Energy from the Scan Mirror and Roof Mirror.**
 - 4a) **Vignetting due to beam translation needs to be characterized.**
 - 2b) **The Scan Mirror is viewed at Normal Incidence. (No additional optics required.)**
 - 3b) **A Silicon Photodiode must be added at the side of the SRCA exit slit to sense the reflected energy from the Scan Mirror.**

Either or both methods can be implemented.
- **Action Needed: If the pursuit of this concept is of interest to GSFC, we'd be happy to respond to a request for proposal.**

SCAN MIRROR REFLECTANCE CHARACTERIZATION

METHOD B - NORMAL INCIDENCE

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